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Technical Report 82099

October 1982

A STUDY OF NORMAL SHOCK-WAVE TURBULENT BOUNDARY-LAYER INTERACTIONS AT MACH NUMBERS OF 1.3, 1.4 AND 1.5

by

W. G. Sawyer Carol J. Long



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Procurement Executive, Ministry of Defence Farnborough, Hants

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SUMMARY

This Report presents the results of a study of seven flows involving the interaction between a normal shock wave and a two-dimensional turbulent boundary layer. The measurements were made at free-stream Mach numbers of 1.3, 1.4 and 1.5 and at Reynolds numbers based on an effective streamwise run of 10×10^6 to 30×10^6 . The results were obtained from comprehensive traverses with both pitot and static probes.

Standard boundary-layer integral parameters based on wall and measured static pressures are presented, together with velocity profiles and the Mach number distribution over the interaction region.

An investigation has been made of the 'law of the wall' and the 'law of the wake' under the influence of strong normal pressure gradients.

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I INTRODUCTION

A series of experiments¹⁻⁴ has been undertaken in the RAE Bedford 3ft × 3ft wind tunnel to investigate the interaction of a normal shock wave and a turbulent boundary layer at nominal upstream Mach numbers of 1.3, 1.4 and 1.5 over a Reynolds number range based on the undisturbed boundary-layer momentum thickness at the start of the interaction of 12×10^3 to 34×10^3 .

This Report deals with the measurements made at a series of stations upstream and downstream of the interaction using conventional pitot and static probes traversed normal to the flat wall.

The series of experiments was planned because of the lack of knowledge of the interaction between normal shock waves and turbulent boundary layers at high Reynolds numbers and the need to predict the development of the boundary layer through and beyond the interaction region as, for example, in the flow over supercritical aerofoils.

A number of investigations have been reported over the past two decades and broadly speaking three techniques have been used to produce the interaction.

The first and most commonly used technique (Fig 1a) in supersonic tunnels is to position a shock generator with downstream choking flap over a flat plate so that the steady normal shock wave formed by the shock generator interacts with a turbulent boundary layer grown from the leading edge of the flat plate. This is the technique adopted by Seddon⁵, Vidal, *et al*⁶ and Kooi⁷.

The second technique^{8,9} is used in transonic wind tunnels and involves mounting a two-dimensional bump on the wind tunnel wall so that the shock-wave boundary-layer interaction approximates to that on an aerofoil (Fig Ib). More recently, small supercritical aerofoils have been tested¹⁰ and in fact these two experiments have been combined by Burdges¹¹ who let the aerofoil into the tunnel floor and bled away the floor boundary layer under its leading edge.

Gadd¹² and more recently Mateer, Brosch and Veigas¹³ used a third technique which was to hold the normal shock wave steady in a supersonic tube by the adjustment of a conical choke downstream of the working section as in Fig Ic. The turbulent boundary layer under consideration was allowed to grow naturally along the tube wall.

For the present experiment the facility which was available was the RAE 3ft × 3ft transonic-supersonic wind tunnel. The arrangements of Fig 1a or 1b could have been employed, but it was argued that if the tunnel floor was used, then the measuring techniques would be simpler, a higher Reynolds number could be obtained, and the interference effects of the wall boundary layers should be less than in the technique of Fig 1a where the test boundary layer is of smaller thickness than the interfering wall boundary layers. The simple technique was therefore used of holding a shock wave across the test section, set up to run at supersonic speed, by means of an adjustable sonic throat which was situated far downstream of the test section. The technique is in fact similar to that of Fig 1c.

The 3ft \leq 3ft wind tunnel has a good Reynolds number capability (up to 12×10^6 /m tor continuous running) is easily accessible for non-intrusive measurements and, above all, is easy to modify it the working section region having removable supersonic liners and downstream wooden fairings. A further advantage is that the pressure distribution in the working section is not only similar to that over a supercritical aerofoil but also the dimensions and Reynolds numbers approximate to full-scale conditions. A ninth scale (4 in \leq 4 in) model of the tunnel is also available and this proved invaluable for the development of the experiment.

The most time-consuming aspect of the development has been providing a suitable downstream sonic throat to control the shock-wave position and to keep it acceptably steady. However it was also necessary to design and manufacture a new raised false tunnel floor (to house submerged boundary-layer traverse mechanisms and to raise the boundarylayer interaction region into view above the bottom of the schlieren windows) which would match the three alternative half-nozzle blocks used to generate the required free-stream Mach numbers and which form the upper wall.

During the period of time taken to manufacture the new floor and the necessary associated equipment, two interim experiments were made, the first² using a conventional boundary-layer pitot rake and the second¹ using non-intrusive laser-Doppler measurements. The first experiment differed in another important aspect from the present one in that the pitot rake was attached to the tunnel floor in a single, fixed, position and the shock wave moved fore-and-aft of the rake.

For the present experiment the location of the shock wave was fixed for each condition tested and measurements were made with probes which could be traversed both normal to the tunnel floor and streamwise. Measurements of the static pressure distribution along the floor were also made.

2 DESCRIPTION OF THE EXPERIMENT

2.1 Mechanical arrangement

The general arrangement of the experiment is shown in Fig 2. The unmodified parts of the $3ft \times 3ft$ wind tunnel are shown shaded.

In its standard form, the 3ft \times 3ft wind tunnel has a working section with a fixed lower liner and interchangeable upper liners which are matched to the lower liner to give 0.1 incremental steps in Mach number between 1.3 and 2.0. One of the major modifications made to the tunnel was the provision of a new raised bottom liner together with a false floor (Fig 3). This false floor contained two pitot traverse mechanisms so that a region close to the floor centre line could be investigated from approximately 0.8 m ahead of the schlieren window centre line to 3 m downstream using a total horizontal traverse movement of little more than 2 m. Because of the interest in the free-stream Mach number range of 1.3 to 1.5, the new bottom liner was designed to match the M = 1.4 upper liner and the consequent slight mismatch with the M = 1.3 and M = 1.5 upper liners had to be accepted. Although raising the floor by 152 mm had the advantage of bringing the shock-wave interaction region into full view through the schlieren windows, it reduced the height of the working section so that the tunnel was no longer square in cross

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section. There was however no particular disadvantage in this because the asymmetrical nozzle which generated the supersonic flow inevitably resulted in boundary layers which were not the same on all four walls.

The other major item of redesign as described in Ref 2 involved the careful fairing of the existing 'spoiler door' arrangement* in the diffuser section to give an adjustable second throat. By adjusting the throat, the normal shock wave could be placed in the desired location wi hin the view through the tunnel windows. Any subsequent small progressive movement of the shock wave could then be corrected by adjusting the tunnel compressor speed. Initially there was a large random movement of the normal shock wave but this was reduced to about 20 mm by the careful fairing of the shape of the second throat. The mean position of the normal shock wave was checked both visually and by means of a differential pressure transducer connected between a wall static pressure hole under the shock wave and a reference tapping near the working section sonic throat.

The region between the working section and diffuser was faired to give smooth sidewall and roof contours. The geometry of the fairing was such as to give an expanding passage, the cross sectional area at the diffuser entry increasing to 1.067 times the area of the working section over a distance of 3660 mm. In the region of the interaction, the standard taper of 0.004 m/m was maintained on the top liner. This is normally combined with a similar taper on the lower liner to allow for boundary-layer growth on all four tunnel walls. However, for this experiment, the taper on the false floor was approximately 0.003 m/m.

The traverse mechanism together with a twin pitot probe is shown in Fig 4. Two identical traverse mechanisms were supported 1800 mm apart from a slide which was approximately 6 m long and sufficiently flexible to submerge, at its upstream end, below the liner surface approximately 1 m ahead of the schlieren window centre line, and at its downstream end to retract into a covered channel between the spoiler doors in the diffuser section. Any leaks around the slide were sealed by two pvc tubes which inflated when the slide was stationary as shown in Fig 3. A chain drive was used to move the slide in the streamwise direction. A total streamwise distance of 3800 mm could be covered using the tandem boundary layer traverses. The slide was positioned 100 mm to the port of the row of static pressure holes on the tunnel centre line and the probes were cranked so that vertical traverses could be made half way between the slide and the tunnel centre line. In this way it was hoped to reduce the various interferences, namely, of the probe on the wall static pressure holes, of the slight irregularity of the slide on the probe measurements, and of the probe stem on the probe measurements.

Repeatability of the probe position was ± 0.02 mm vertically and ± 3 mm horizontally.

Details of the static probes can be seen in Fig 5. The pitot probes were designed to be fairly short in order to avoid vibrations in the expected highly turbulent flow. The same overall dimensions were then retained for the twin static probes. With hindsight, the twin static probes might have been lengthened to diminish the interference

* Used in normal testing for controlling the tunnel shock wave during starting or stopping the tunnel.

caused by the supporting structure because the vertical positional accuracy is not quite as important for these probes as for the pitot probes. However this was not done and large corrections had to be applied to the static pressure measurements especially at transonic Mach numbers. These corrections are described in the Appendix.

All the probes were electrically insulated from their vertical traverse mechanisms so that a touch indicator could be used to set the pitot datum at the surface of the false floor. The streamwise datum was set by aligning the tip of a pitot probe or the holes in a static probe with a line marked on the surface of the false floor.

Pressure measurements were made with differential transducers (Druck) of range $\cdot 69 \text{ kN/m}^2$ in two D-type scanivalves placed outside the tunnel shell. The transducers were calibrated against a Texas Instrument 0-203 kN/m² absolute quartz Bourdon-tube pressure-controller. Corrections were made to the primary slopes and zeros of these calibrations for each data point, by comparing the transducer outputs with reference pressures applied to the first and last pairs of scanivalve ports.

2.2 Experimental measurements

2.2.1 Pitot measurements

Measurements were made at three nominal Mach numbers, 1.3, 1.4 and 1.5 at Reynolds numbers of approximately $10 \times 10^6/m$ (the maximum continuous running value permitted for the experiment) and $3.5 \times 10^6/m$. An intermediate Reynolds number was also included for M = 1.5. Details of the test conditions are given in Table 1.

Traverses normal to the floor were made using the twin-pitot probe over a large number of stations f om 0.7 m ahead of the normal shock wave to approximately 3 m downstream of the normal shock wave. In most cases, a vertical distance of 200 mm was covered. The forward pitot was removed while measurements were being made with the rear one, and a flush plug was used to fill the hole left in the slide.

2.2.2 Floor static pressure measurements

Measurements of the static pressure distribution on the floor were made with only the rear pitot in place, set at its furthest downstream position. These measurements were used to calculate the wall reference static pressure for each pitot or static pressure probe traverse.

2.2.3 Reversed pitot measurements

Measurements were also made with a single reversed pitot probe (rather similar in shape to static probe B (Fig 5) but reversed) for conditions of separated flow which occurred for M = 1.5. Separation was indicated by the forward-pointing pitot recording a lower pressure than the corresponding wall static pressure hole. The measurements were made with the tip of the reversed pitot in the same streamwise position as the normal pitot tube. Although the flow was probably disturbed by either probe, a simple correction was made to both sets of results as described under section 3.1.

2.2.4 Static probe measurements

Before the static pressure measurements in the boundary layer were begun, the static probes were calibrated in the centre of the slotted transonic working section of

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the tunnel over a Mach number range of 0.4 to 1.2 and similar free-stream Reynolds numbers to those encountered in the main experiment. Each probe was held at the end of its stem furthest away from the probe tip by a specially manufactured sting. Static probe B was also calibrated while supported at the near end of its stem to check for support interference. This appeared to be negligible. The probe measurements were compared with wall static pressures obtained from a tunnel calibration referred to a 3° conical static head probe. This probe has been described in Ref 14 and within the limits of the required accuracy was assumed to have no static pressure errors.

The real problem which arose when using static probes in an environment which invoked the problems \odot , transonic flow combined with pressure gradients, wall interference effects and the presence of boundary layers, lay in obtaining sufficient calibration data in a tunnel which could not be used between M = 0.88 and M = 1.3 while modified to take the false floor and traverse mechanisms.

However some information was forthcoming from twin static probe calibrations made through the floor boundary layer between M = 0.66 and 0.88 at similar Reynolds numbers to the main experiment. Additional information was obtained during the actual shock-wave boundary-layer interaction experiment from the static and pitot probe traverses ahead of the interaction, while further data were available from traverses 1 to 3 m downstream of the normal shock wave at M = 1.3. In all these cases it was assumed that there was no static pressure variation normal to the wall through each vertical boundary-layer traverse, and that the pressure was equal to the estimated wall value. Because the freestream calibration showed no apparent Reynolds number effects, data at the highest Reynolds number from the experiment were used.

Time limitations meant that nearly all the boundary layer static traverses were made with the twin-static probe rather than the single probes and the static investigation was omitted for M = 1.4 at a Reynolds number of 3.5×10^6 /m. Single static probes were in fact used only as a check on the validity of the results obtained from the twin probe. Traverses were made in rather different streamwise positions from those made with the pitot probes and each traverse contained about half the measurements made using the pitot.

2.2.5 Shock-wave position

A careful check was made of the shock-wave position using the output from the shock-position transducer and readings were only taken while the shock wave was within prescribed limits. Subsequently, during the analysis of the results, the average position of the normal part of the shock wave was ascertained from the large number of schlieren photographs taken for each set of tunnel conditions. This was necessary because there was an interval of a year between making the static pressure measurements and the pitot measurements. It was difficult to reset the shock wave in the same position, and so corrections had to be applied to make both sets of results compatible.

3 REDUCTION OF DATA

The data have been reduced as follows to produce the boundary layer profiles of Table 2 and the integral parameters of Table 3.

3.1 Boundary-layer profiles

These have been tabulated against y (the equivalent height of the pitot or static tube above the wall) where for pitot measurements

$$y = h + 0.15 d$$
 (1)

and for static measurements

$$\mathbf{y} = \mathbf{h} \tag{2}$$

where h = height of probe centre line above floor and d = diameter of probe.

The profiles have been derived

- (a) assuming constant wall static pressure applied throughout each traverse,
- (b) using measured static pressures after correction (see Appendix).

Standard formulae were used to calculate Mach numbers, velocities and densities from the pitot and static results. Total temperature was assumed constant across the boundary layer except at the wall where a recovery factor of 0.89 was used. Wall static pressures were calculated for each traverse position by linear interpolation of the floor static measurements made with the front pitot removed and the rear pitot as far downstream as possible.

The flow was assumed reversed when the pitot pressure was lower than the wall static pressure. In such cases measurements were also made with the pitot tube reversed in direction. The Mach number was taken as the mean derived from the two sets of measurements. When the pitot tube was effectively facing downstream, it was assumed that the pressure recorded was that for a base with a pressure coefficient of -0.6. The wall static pressure was used for both sets of calculations in the reversed flow region.

Unit Reynolds numbers were calculated using the following formula based on Sutherland's law for viscosity

$$\operatorname{Re/m} = \frac{M_{\delta} p_{\delta}}{T_{\delta}^{2}} (T_{\delta} + 110.4) \times 47.91 \times 10^{3}$$
(3)

where subscript δ denotes conditions at the edge of the boundary layer

p is the static pressure in N/m^2 and T is the temperature in K.

A further parameter, namely p_i/p_{t_0} has been presented where p_i is an estimated static pressure in the 'equivalent inviscid flow' (see Ref 15). The static pressure p_i is that in the equivalent inviscid flow which is defined as the flow external to the shear layers continued as a: inviscid flow to the wall bounding the real flow with the growth of the viscous layer represented by transpiration at the boundary. p_i has been non-dimensionalised by dividing by the tunnel total pressure p_{t_0} .

An estimate of p_i/p_{t_0} has been attempted in the region ahead of the normal shockwave. Here, the flow at the edge of the boundary layer is supersonic and influence lines of constant p/p_{t_0} following Mach lines may be constructed back to the wall from the edge of the viscous region. An estimate of p_i/p_{t_0} as a function of y for each traverse can then be made from the carpet of influence lines which are assumed straight and inclined at the Mach angle μ plus the flow inclination at $\delta_{0.999}$ (the distance from the wall where $U/U_{\delta} = 0.999$) viz

$$\mu + \tan^{-1}\left(\frac{v}{U}\right)_{\delta}$$

where

$$\left(\frac{V}{U}\right)_{\delta} = \frac{d\delta^{\star}}{dX} - (\delta_{0.999} - \delta^{\star}) \frac{d}{dX} \left(\ln(\rho_{\delta}U_{\delta})\right)$$
(4)

and V is the vertical component of velocity

- U is the horizontal component of velocity
- δ^* is the displacement thickness
- ρ is the density.

It will be noted (see section 4.4) that there is a supersonic region at the edge of the boundary layer behind the normal shock wave for M = 1.5. However estimates of p_i/p_t have not been included for this region because they become discontinuous in the region of the shock wave.

3.2 Integral parameters

Boundary-layer thickness parameters were obtained by trapezoidal integration. Velocity profiles were faired between the wall and the pitot position corresponding to 0.8 mm from the wall by applying East's prediction of the law of the wall¹⁶ in compressible boundary layers as a 20 point curve.

East's prediction is based on an incompressible law of the wall combined with a compressibility factor. For this experiment Cole's law of the wall¹⁷ has been used with the constants recommended at the Stanford conference¹⁸ namely

$$\frac{U}{U_{\tau}} = 5.62 \log \frac{yU_{\tau}}{v_{\delta}} + 5.0 , \qquad (5)$$

where ${\rm U}_{\tau}$ is the friction velocity.

The form of East's prediction which is faired into Cole's law of the wall is

$$\frac{U}{U_{\tau}} = \frac{1}{F} \sin \left[F \left\{ 2K \ln \left(y^{\star 2} \frac{K}{D} + 1 \right) + D \left(1 - e^{-y^{\star}/D} \right) \right\} \right] , \qquad (6)$$

where the compressibility factor

$$F = M_{\tau} \sqrt{\frac{r(\gamma - 1)}{2}} = \frac{U_{\tau}}{\sqrt{T_{w}}} \times 0.021033 ,$$

$$D = 8.73(1 + 45F^{2}) ,$$

$$y^{\star} = y \frac{U_{\tau}}{v_{w}} ,$$

$$K = 0.41 ,$$

r being the recovery factor, and suffix w denoting wall values.

A value of U_{τ} was obtained at $y \simeq 0.8 \text{ mm}$ using equation (6) and values of U/U_{τ} were then calculated for 20 equispaced values of y between 0 and 0.8 mm.

A linear variation was assumed between y = 0 and 0.8 mm for reversed flow.

The following definitions of the integral quantities have been used as first proposed by Myring 19 and later incorporated in East's modified momentum integral equation 15 .

They are (where suffices i and w denote equivalent inviscid flow quantities and wall values respectively)

$$\overline{\delta} = \frac{1}{\rho_{iw} U_{iw}} \int_{0}^{\delta} (\rho_{i} U_{i} - \rho U) dy , \qquad (7)$$

$$\tilde{\theta} = \frac{1}{\rho_{iw} U_{iw}^2} \int_0^{\infty} \left\{ \left(\rho_i U_i^2 - \rho U_i^2 \right) - U_{iw} \left(\rho_i U_i - \rho U \right) \right\} dy , \qquad (8)$$

shape parameter $H = \frac{\overline{\delta}}{\overline{\theta}}$, (9)

shape parameter
$$\tilde{H} = \frac{1}{\bar{\theta}_{\rho}} \int_{iw}^{\rho} (U_i - U) dy$$
 (10)

The integral parameters were calculated for three different distributions of static pressure across the boundary layers as described in the following paragraphs.

3.2.1 Static pressure constant and equal to the measured wall pressure

Equations (7), (8), (9) and (10) reduce to the familiar standard integrals when static pressure is constant across the boundary layer giving

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displacement thickness
$$\delta^* = \int_0^{\delta} \left(1 - \frac{\rho U}{\rho_0 U_{\delta}}\right) dy$$
, (1)

momentum thickness

$$0 = \int_{0}^{0} \frac{\rho U}{\rho_{0} U_{\beta}} \left(1 - \frac{U}{U_{\beta}}\right) dy , \qquad (12)$$

shape factor
$$H = \frac{\delta^*}{\delta}$$
, (13)

shape factor
$$\overline{H} = \frac{1}{2} \int_{U} \frac{\omega}{\omega_{\delta}} \left(1 - \frac{U}{U_{\delta}} \right) dy$$
 (14)

The energy thickness was also evaluated:
$$\delta_{\mathbf{E}} = \int_{0}^{\delta} \frac{c U}{c_{\beta} U_{\beta}} \left[1 - \left(\frac{U}{U_{\beta}} \right)^{2} \right] dy$$
 (15)

3.2.2 Static pressure variation as measured

Equations (7), (8), (9) and (10) can be used in an 'intermediate form' as in Cook. McDonald and Firmin²⁰. In this form the boundary layer defect thicknesses are reformed to a fictitious potential flow having the same static pressure distribution as that for the actual viscous flow and a total pressure equal to that at the edge of the boundary layer (p_{t_1}) .

Thus for example

$$M_{p} = \sqrt{5 \left[\left(\frac{p_{t_{1}}}{p} \right)^{\frac{2}{7}} - 1 \right]}$$
(16)
$$p_{t_{1}} = p_{\delta} \left(1 + 0.2 M_{\delta}^{2} \right)^{3.5} .$$
(17)

where

The revised integral quantities are as follows:

displacement thickness
$$\delta^* = \frac{1}{\rho_w U_w} \int_0^{\delta} (\rho_p U_p - \rho U) dy$$
, (18)

momentum thickness
$$\theta = \frac{1}{\rho_w U_w^2} \int_0^{\delta} \left\{ \left(\rho_p U_p^2 - \rho U^2 \right) - U_p \left(\rho_p U_p - \rho U \right) \right\} dy , \quad (19)$$

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(17)

shape factor
$$H = \frac{\delta \star}{\Theta}$$
,

shape factor
$$\overline{H} = \frac{1}{\theta} \frac{1}{\rho_{W} U} \int_{0}^{\infty} \rho_{p} (U_{p} - U) dy$$
 (20)

3.2.3 Equivalent inviscid static pressure together with measured static pressure

 $\overline{\delta}$, $\overline{\theta}$, H and \overline{H} have already been defined by equations (7), (8), (9) and (10).

Equivalent inviscid static pressures were estimated as described in section 3.1 for traverse positions ahead of the normal shock. Equivalent inviscid Mach numbers were obtained from these pressures, and the total pressure calculated at the edge of the boundary layer (equation (17)) so that

$$M_{i} = \sqrt{5 \left[\left(\frac{p_{t_{i}}}{p_{i}} \right)^{2} - 1 \right]}$$
 (21)

 c_i and U_i were obtained from M_i using a temperature recovery factor of unity while the values of ρ and U were identical to those in section 3.2.2 (being calculated from the measured pitot and static pressures).

Linear interpolation was used to obtain ρ_i and U_i for the 20 points between the wall and y=0.8~mm .

3.3 Skin friction

Three estimates of skin friction are given.

The first estimate C_{fp} has been deduced by considering the pitot tube when in contact with the wall to be a Preston tube and applying Patel's²¹ calibration as formulated by Head and Vasanta Ram²² and transformed for compressible flow by the method of Fenter and Stalmach²³.

The second and third estimates emerge directly from U_{τ} which was calculated by fitting East's law of the wall (equation (6)) to the pitot reading for $y \simeq 0.8$ mm. The second estimate was obtained assuming constant wall static pressure to apply across the boundary layer and the third was obtained using measured static pressures. Because at $y \simeq 0.8$ mm wall static pressure applied for both cases, the differences are entirely due to the boundary layer edge conditions obtained from the different static pressures.

It will be noted that several values of C_f are missing from Table 3 and these omissions occur where reversed flow is indicated at M = 1.5.

4 EXPERIMENTAL RESULTS

4.1 The general characteristics of the flow

4.1.1 Shock-wave patterns

Fig 6a shows the vertical cross sections of the main interactions traced from schlieren photographs. Typical photographs have been reproduced in Fig 6b. For completeness, the calculated boundary layer thickness ($\delta_{0.995}$) and the start of the measured reversed flow regions at M = 1.5 have been added to the diagrams. The progression from the pattern at M = 1.3 of a single normal shock wave distorted by the influence of compression curves emanating from the thickening of the upstream boundary layer, to the established triple shock system with the shear layer emanating from the point of bifurcation at M = 1.5, is well illustrated. The increase in size of the shock system with reduction in Reynolds number can also be seen.

The main interaction region has been described in detail by East³ and the present results at high Reynolds numbers agree with his at M = 1.3 and 1.4. However there is a discrepancy between the sets of results at M = 1.5. The shock bifurcation point for the present experiments is at 180 mm from the floor compared with a value, presumed to be erroneous, of 215 mm given by East.

4.1.2 Flow visualisation

After the completion of the traverse measurements the interaction region was examined by oil flow*.

Photographs of the surface oil flow were taken for all three Mach numbers at a Reynolds number of 10×10^6 /m. These photographs, reproduced in Fig 7, were taken after shutting down the tunnel and removing the roof and therefore suffer from slight blurring. The viewpoint looks downstream. The photographs reveal mild three-dimensional effects for M = 1.3 and M = 1.4 but a much more complex pattern for M = 1.5 when the flow is separated. The photograph for M = 1.5 is shown in more detail in Fig 8 together with an attempted interpretation of the wall streamlines.

The nature of the flow in this case appears to be such that there is flow along the arch of a vortex connecting a position on the port side, denoted by B in Fig 8, to a position denoted by A on the starboard side. Thus in a sense A is an attachment node and B a separation node. It should be remarked that whilst the streamwise extent of this region is roughly 300 mm the probe measurements suggest its depth to be only 7 mm.

Near the tunnel centre line between the two vortex patterns, the flow is tolerably two-dimensional except in the immediate regions of the saddle points at the start and end of the separation region.

Presumably because of the non-uniform boundary-layer thickness on the tunnel side wall arising from the flow field associated with the assymetrical nozzle, the separation line on the sidewall is swept and a single vortex node only of separation type is formed.

^{*} The oil-flow mixture was an amalgam of the following in the ratio of 4 cc Vitrea F2, to 2 cc Limea 931, to 3 cc TiO₂, to 2 drops of oleic acid.

The vortex arising from this node is presumed to create the strong convergence towards the tunnel centre line occurring in the corner regions downstream of the interaction. The lines D in Fig 8 have the appearance of separation lines but may in fact indicate only a locally strong convergence of the boundary layer.

4.1.3 Momentum balance

Momentum balance calculations have been made using a rearranged form of East's wodified momentum integral equation 15

$$\frac{d\overline{\vartheta}}{dx} + \left(H + 2 - M_{iw}^{2}\right) \frac{\overline{\vartheta}}{U_{iw}} \frac{dU_{iw}}{dx} - \frac{\tau_{w}}{\rho_{iw}U_{iw}^{2}}$$

$$= -\frac{1}{2c_{iw}U_{iw}^{2}} \frac{d}{dx} \left\{ \rho_{iw}U_{iw}^{2} \left(\frac{d^{2}\overline{\delta}}{dx^{2}} + \kappa \right) (\overline{\vartheta} + \overline{\delta})^{2} \right\} + \kappa (\overline{\vartheta} + \overline{\delta}) \frac{d\overline{\delta}}{dx}$$

$$+ \frac{1}{\rho_{iw}U_{iw}^{2}} \frac{d}{dx} \left\{ \kappa \rho_{iw}U_{iw}^{2} \overline{\vartheta} \left(\frac{\overline{H} - 1}{\overline{H}} \right) \right\} + \frac{M_{iw}^{2} \overline{\vartheta} V_{iw}}{U_{iw}^{2}} \frac{dV_{iw}}{d\overline{X}} . \qquad (22)$$

The last term has been neglected in the analysis because it is small and in any case cannot be evaluated with any reliability. The remaining terms in the right hand side of the equation are:

the normal pressure gradient effect arising from the wall and stream curvature;
 the approximation to the direct effect of wall curvature on the flow momentum;
 the approximation to the normal stress terms.

The integral parameters, H, \tilde{H} , $\bar{\theta}$ and $\bar{\delta}$ are defined in equations (7) to (10) while ρ_{iw} , U_{iw} and M_{iw} are respectively the equivalent inviscid values of density, horizontal component of velocity and Mach number at the wall.

The surface curvature κ is zero for this example leaving

$$\frac{d\overline{\theta}}{dx} + \left(H + 2 - M_{iw}^{2}\right) \frac{\overline{\theta}}{U_{iw}} \frac{dU_{iw}}{dx} - \frac{\tau_{w}}{\rho_{iw}U_{iw}^{2}}$$

$$= -\frac{1}{2\rho_{iw}U_{iw}^{2}} \frac{d}{dx} \left\{\rho_{iw}U_{iw}^{2} \frac{d^{2}\overline{\delta}}{dx^{2}} \left(\overline{\theta} + \overline{\delta}\right)^{2}\right\} + \frac{1}{\rho_{iw}U_{iw}^{2}} \frac{d}{dx} \left\{Kc_{iw}U_{iw}^{2}\overline{\theta}\left(\frac{\overline{H} - 1}{\overline{H}}\right)\right\}. \quad (23)$$

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After multiplying throughout by $arphi_{iw}^2$ and rearranging, the equation becomes

$$\frac{d}{dX} \left(\rho_{iw} U_{iw}^2 \overline{\theta} \right) = \tau_w - H \rho_{iw} U_{iw} \overline{\theta} \frac{dU_{iw}}{dX} + \frac{d}{dX} \left(f \rho_{iw} U_{iw}^2 \overline{\theta} \right) , \qquad (24)$$

where $f = f_1 - f_2$, $f_1 = 0.072\left(\frac{\overline{H} - 1}{\overline{H}}\right)$, and is the contribution due to the normal stress, and $f_2 = \frac{1}{2} \frac{d^2 \overline{\delta}}{dx^2} \frac{1}{\overline{H}} (\overline{H} + \delta^2) = \frac{1}{2} \frac{d^2 \overline{\delta}}{dx^2} (1 + H) (\overline{\theta} + \overline{\delta})$ and is the contribution due to the stream curvature

If suffix 0 represents the conditions at the start of the interaction, then dividing through by $\rho_{iw_0} v_{iw_0}^2 \overline{\theta}_{iw_0}$ and integrating, results in the form of the momentum integral equation used in the momentum balance calculations shown in Fig 9:

$$\begin{bmatrix} \frac{\rho_{iw}U_{iw}^{2}\overline{\theta}}{\rho_{iw}U_{iw}^{2}\overline{\theta}_{0}} - 1 \end{bmatrix} = \int_{X_{0}}^{X} \frac{c_{f}}{2} \frac{\rho_{e}U_{e}^{2}}{\rho_{iw}U_{iw}^{2}\overline{\theta}_{0}} dX - \frac{1}{2} \int_{I}^{P_{iw}\overline{\delta}} \frac{\rho_{iw}\overline{\delta}}{\rho_{iw}\overline{\delta}_{0}} d\left(\frac{U_{iw}}{U_{iw}}\right)^{2} + \left(\frac{f\rho_{iw}U_{iw}^{2}\overline{\theta}}{\rho_{iw}U_{iw}^{2}\overline{\theta}_{0}} - f_{0}\right)$$

..... (25)

In Figs 9 and 11 $\left(\frac{\rho_{iw}U_{iw}^2\overline{\theta}}{\rho_{iw_0}U_{iw_0}^2\overline{\theta}_0} - 1\right)$ is called the left hand side while the remainder

of the expression is called the right hand side. The numbered terms are identified later (in Fig 12) as the contributions due to

5 pressure gradient.

Both sides of equation (25) were calculated using the alternative assumptions of section 3.2 which were (for traverses across the boundary layer):

- (a) static pressure constant and equal to the measured wall pressure;
- (b) static pressure as measured;
- (c) equivalent inviscid static pressure together with measured static pressure. The equivalent inviscid static pressure was calculated for traverses ahead of the shock wave only.

Fig 9 shows the comparison between the left hand side and the right hand side of equation (25) plotted against the streamwise position. The calculations were made using assumption (c). It will be seen that the momentum balance is good over the main interaction region and only becomes significantly in error at approximately 800 mm downstream of the normal shock wave. Downstream of this point, the left hand side actually reduces f_{σ^l} Mach numbers ahead of the interaction of 1.4 and 1.5. This indicates a flow divergence and it is interesting to calculate its magnitude.

In terms of equation (22) Green *et al*²⁴ showed that a flow divergence could be accounted for by subtracting the local rate of divergence $\overline{\theta}(d\phi/dz)$ from the right hand side. This quantity represents the rate of the deviation relative to the nominal flow direction with respect to distance Z perpendicular to the flow but parallel to the surface.

In equation (25) this divergence takes the form

$$\int_{X_0}^{X} \frac{\rho_{iw} U_{iw}^2 \overline{\theta}}{\rho_{iw_0} U_{iw_0}^2 \overline{\theta}_0} \frac{d\phi}{dz} dx$$

and this can be added to the left hand side. It is therefore easy to make an estimate of $d\sigma/dz$.

This estimate is shown in Fig 10a-c in degrees per metre. For points further downstream of the normal shock wave than 300 mm the figures show a flow divergence of $7^{\circ}/m$ at M = 1.3 rising to $15^{\circ}/m$ at M = 1.5 (equivalent to 7° and 15° over the tunnel width). The wild fluctuations within the main interaction region are of course due to the rapidly changing conditions and the close streamwise spacing of the traverse positions.

Assumptions (a), (b) and (c) were used to provide comparisons of both sides of the momentum integral equation as shown in Fig 11. There appears little to choose between the momentum balance using any of these assumptions. However it is possible to see some improvement using assumption (b) rather than (a). If the region ahead of the shock wave is examined there is again some improvement if calculation assumption (c) is used.

Fig 12 shows the individual contributions to the right hand side of equation (25).

4.2 Edge Mach numbers and boundary layer parameters

Fig 13a-c shows the development of the Mach number at the edge of the boundary layer, M_{δ} , the boundary-layer displacement thickness, δ^* , the shape factor H, defined by equation (13), the boundary layer thickness, $\delta_{0.995}$, the skin friction coefficient C_{f_p} (section 3.3), and the shape factor \tilde{H} defined by equation (14), all plotted against the longitudinal distance X from the normal part of the shock wave. The results in Fig 13 were calculated assuming constant static pressure, p_w , across the boundary layer normal to the tunnel wall. Further graphs of M_{δ} , δ^* , H and \tilde{H} (Fig 14a-c) show the effect of using measured static pressures to calculate these parameters. The longitudinal region covered by these figures is from 500 mm ahead to 500 mm downstream of the normal shock wave. Outside this region it was assumed that the static pressures were constant normal to the wall and equal to the static pressures used in the first set of calculations (see Fig 15a&b and the Appendix).

Overall there appears to be little effect of variation of Reynolds number on the shape factors and the normalised variation of δ^* and $\delta_{0.995}$. A separation bubble occurs under the interaction region at M = 1.5 and its extent is indicated by the zero values of C_{f_p} . The bubble size decreases slightly with increase in Reynolds number. The shape of the bubble can, however, be seen in the plots of Mach number distribution, Fig 16a-f, and will be described in section 4.4.

The results agree broadly with those obtained in Ref 2 where a fixed pitot rake was used on the standard tunnel floor. However discrepancies begin to creep in 1 m downstream of the normal shock wave where in Ref 2 the Mach number at the start of the interaction was affected by the proximity of the tunnel throat. The results from Ref 2 at M = 1.5were much more sensitive to Reynolds number in the interaction region and this was probably due to pitot-rake interference. One other point should be noted (and it can be seen in nearly all the following figures) and that is that the rehabilitation process behind the normal shock wave has not been completed after 3 m. This distance is equivalent to at least 400 times the undisturbed displacement thickness ahead of the interaction. It can be seen in Fig 13a-c that the edge Mach numbers, shape factors and boundary layer thicknesses have not yet reached their asymptotic values at the furthest downstream traverse positions.

4.3 Measured static pressures

Static pressure distributions normal to the wall are plotted on scale diagrams of the main interaction region in Fig 15a&b as the difference from the corresponding wall static pressure non-dimensionalised by dividing by the undisturbed stagnation pressure. Zero values are located at the appropriate streamwise station.

As described in the Appendix, the static pressure results have usually been made to merge into the appropriate wall values at floor level, however wall static values have been assumed to apply throughout the separated region in the absence of reliable local values. Static pressures recorded very close to the shock wave may have been affected by its movement and so local pressure gradients may have been reduced.

The diagrams show the expected static pressure variation throughout the main interaction region. The static pressures are fairly constant across the inner part of the shear flow and then decrease outwards across the compression region ahead of the normal shock wave to free-stream values. Behind the main shock wave they increase from one fairly constant level near the wall to a second level behind the normal part of the shock wave.

4.4 Mach number distribution

Mach number contours are plotted in Fig 16a-f over the main interaction region and over the whole of the region investigated downstream of the normal shock wave (note the change of streamwise scale). Measured pitot and the corrected static pressures were used to calculate Mach numbers and the figures result from linear interpolations of Mach number profiles in the vertical direction and from horizontal plots of Mach numbers at constant distances from the wall, y.

The results are therefore dependent on the accuracy of the corrected static pressure measurements and the interpolation procedures which are affected most critically by the smearing effect of the slightly unsteady shock wave. However the results in the interaction region are broadly confirmed by the laser-Doppler anemometer measurements of East³ and the results of Kooi⁷ at M = 1.4. A particular feature is the development of the supersonic tongue beneath the downstream shock wave for M = 1.5.

If East's and the present results at a Reynolds number of 10×10^{6} /m are compared in slightly more detail, the Mach number contours are in good agreement ahead of the normal shock wave if due allowance is made for the smearing effect of the unsteady shock wave (see above). Very similar supersonic tongues may be seen behind the shock wave and they continue downstream (in both experiments) for 100 mm near the edge of the boundary layer. East's results at M = 1.5 do not cover the full extent of the supersonic tongue but show that it is more extensive than at M = 1.4. In fact the present experiment confirms that it persists near the edge of the boundary layer for 400 mm downstream of the shock wave while at a Reynolds number of 3.5×10^{6} /m it persists for 600 mm.

It is difficult to be specific about the shape of the supersonic tongue immediately downstream of the trailing shock wave, but the present results agree well with Kooi⁷ who also used pitot and static pressure measurements. The rear shock wave appears to terminate at the edge of the boundary layer so that supersonic flow appears to exist above the boundary layer behind a normal shock wave.

There is evidence to suggest that the thin shear layer formed downstream of the shock bifurcation point is still intact 3 m behind the normal shock wave, and even at M = 1.3 there is no sign that the inviscid flow has recovered sufficiently to provide a uniform Mach number contour across the tunnel.

The other point of interest is the separation region at M = 1.5. There appears to be a small decrease in the overall dimensions of the separation bubble with increase in Reynolds number. In fact the length of the separation bubble is reduced from 300 mm at $\text{Re/m} = 3.5 \times 10^6$ to 200 mm at $\text{Re/m} = 10 \times 10^6$ while the height is reduced slightly from just over 9 mm at the lowest Reynolds number to 7 mm at the highest Reynolds number. This result is in disagreement with Sawyer *et* at^2 where a much greater reduction in bubble size was indicated although as stated in section 4.2 the conditions were somewhat different.

4.5 Boundary-layer velocity profiles

4.5.1 General

Boundary-layer profiles are shown in Fig 17a-f plotted on a scale diagram of the main interaction region together with an extended diagram on a compressed scale showing the whole region of measurement behind the normal shock wave.

The figures indicate the general characteristics of the boundary-layer development. They show the progression from the undisturbed state through the leading compression which influences the shape of the outer part of the boundary layer, through the subsequent separation or near-separation region, to the final recovery towards a zero-pressuregradient form for the profiles. It can be seen even on the small scale diagrams that the profiles are still disturbed after a distance of nearly 3 m downstream of the normal shock wave, or approximately 400 times the undisturbed displacement thickness.

4.5.2 Logarithmic velocity profiles

Representative velocity profiles are plotted in the logarithmic form adopted by Winter and Gaudet²⁵ as $\left(U/U_{\tau}^{i}\right)$ against $\log(yU_{\tau}^{i}/v_{\delta})$ where U_{τ}^{i} is the equivalent

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incompressible friction velocity. Values of U_{τ}^{i} were derived from the skin friction coefficient $C_{f_{\tau}}$ using the formula

$$U_{\tau}^{i} = U_{\delta} \left(\frac{C_{fp}}{2}\right)^{i} \left(1 + 0.2M_{\delta}^{2}\right)^{i} \qquad (26)$$

 Cf_p was obtained by treating the pitot tube when in contact with the wall as a Preston tube and using the calculation method described in section 3.3.

The results are plotted separately for three regions. Fig 18 shows the undisturbed velocity profiles just ahead of the interaction region. These profiles were used to define the undisturbed conditions of Table 1. The profiles far downstream of the interaction region are shown in Fig 19. Fig 20 shows typical profiles just downstream of the main interaction region.

4.5.3 The law of the wall

Figs 18 to 20 show typical 'law of the wall' fits for small values of y (the distance from the wall). The law of the wall is defined in the form adopted by Winter and $Gaudet^{25}$ as

$$\frac{U}{U_{\tau}^{i}} = \frac{1}{\kappa} \ln\left(\frac{yU_{\tau}^{i}}{v_{\xi}}\right) + \phi(0) \quad , \qquad (27)$$

which may be expressed as

$$\frac{U}{U_{\tau}^{i}} = A \log\left(\frac{yU_{\tau}^{i}}{v_{\delta}}\right) + B \quad .$$
 (28)

The results have been compared with Winter and Gaudet's incompressible law of the wall

$$\frac{U}{v_{\tau}^{1}} = 6.05 \log\left(\frac{yU_{\tau}^{1}}{v_{\delta}}\right) + 4.05 , \qquad (29)$$

and with Coles' incompressible form (equation (5))

$$\frac{U}{U_{\tau}} = 5.62 \log\left(\frac{yU_{\tau}}{v_{\delta}}\right) + 5.0$$

Coles' incompressible law of the wall is usually applied to compressible flow by making use of the Van Driest²⁶ transformation. It was therefore necessary to confirm that the transformations of Van Driest and Winter and Gaudet gave similar results when applied to the compressible profiles of the present experiment. Two profiles were checked and are plotted using both transformations in Figs 18 and 19. The first profile had a free-stream Mach number of 1.54 and was in the undisturbed region ahead of the

interaction region, while the second profile had a free-stream Mach number of 0.83 and was from a pitot traverse made 2759 mm downstream of the normal shock wave. It can be seen that both transformations gave identical values of A and B although the wake components were very different at the free-stream Mach number of 1.54 (Fig 18). Because both transformations gave identical values of A and B Coles' incompressible law of the wall has been presented in the form adopted by Winter and Gaudet.

It would appear from Fig 18 that within the data scatter there is little to choose between the forms of the law of the wall due to Coles and due to Winter and Gaudet.

The average fit made to all the profiles ahead of the interaction region is

$$\frac{U}{U_{1}^{i}} = 6.16 \log \frac{y U_{\tau}^{i}}{v_{\delta}} + 3.6$$
(30)

which is somewhat closer to the latter.

However behind the normal shock wave (Figs 19 and 20) the average fit becomes

$$\frac{U}{U_{\tau}^{i}} = 4.77 \log \frac{y U_{\tau}^{i}}{v_{\delta}} + 6.6$$
(31)

which is quite different from equation (30).

Equations (30) and (31) were obtained by fitting the best straight lines (using the method of least squares) to the linear logarithmic regions close to the wall and averaging the results ahead of, and downstream of, the normal shock wave. Straight line fitting was not attempted when there were less than eight measured points in the linear region, or when the pitot traverse results contained reversed flow. The two traverse positions that were closest to the wall were omitted when fitting best straight lines.

Fig 21 shows the variation of A and B with streamwise position for each undisturbed free stream condition. It demonstrates most clearly the reduction in A accompanied by an increase in B for the boundary-layer traverse positions downstream of the normal shock wave. There does not seem to be any consistent variation of A and B with Reynolds number.

It would appear therefore that either large perturbations or normal pressure gradients affect the law of the wall, in a way which implies an increase in the eddy viscosity.

4.5.4 Departures from equilibrium and the law of the wake

An attempt has been made to demonstrate violent departures of the flow from equilibrium by extracting from the measurements the parameters used to describe the equilibrium locus in integral models of turbulent boundary layers. The analysis is further extended in terms of the character and magnitude of the wake component.

In order to deal with separated flows, East, Smith and Merryman²⁷ redefined the usual parameters

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$$G = \frac{\overline{H} - 1}{\overline{H}} \sqrt{\frac{2}{C_f}} , \qquad (32)$$

$$\Pi = -\frac{2\delta^{\star}}{C_{f}U_{\delta}}\frac{dU_{\delta}}{dx}$$
(33)

$$E_{f}^{i} = \frac{1}{G^{2}} , \qquad (34)$$

and

as

$$E_{p}^{i} = \frac{\Pi}{G^{2}} , \qquad (35)$$

thus removing the difficulty created as the skin friction coefficient passes through zero and becomes negative. With an empirically derived allowance for compressibility, the parameters become

$$E_{f} = \frac{1 + 0.04M_{\delta}^{2}}{G^{2}}$$
(36)

$$E_{p} = \frac{\Pi}{G^{2}} \left(1 + 0.04 M_{\delta}^{2} \right) .$$
 (37)

In Fig 22, the loci for the measured boundary layer developments are compared with the equilibrium locus

$$E_{f} = 0.024 - 0.8E_{p}$$
 (38)

which is substantiated by experiment for attached flow and tentatively assumed in Ref 27 to remain valid in separated flow.

Points to the right and left of the equilibrium locus correspond respectively to stronger and weaker adverse pressure gradients than those appropriate to equilibrium flow. Negative values of E_p represent favourable pressure gradients, while negative values of E_r represent separation.

In Fig 22 the results for each set of test conditions have been plotted using symbols to represent the following regions

- A represents equilibrium conditions ahead of the main interaction;
- B represents the strongly adverse pressure gradient under the leading compression (which causes separation at M = 1.5);
- C represents the rapid recovery just downstream of the normal shock wave;
- D represents the remainder of the flow which might be expected to approach equilibrium conditions.

Points on the loci where there are changes from one region to the next are indicated by the numerals 1 to 3.

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The loci show that region A is approximately in equilibrium but the rapid rise of E_p while E_f only falls slowly in region B indicates that the boundary layer lags in its response to the strong adverse pressure gradient under the leading compression. In region C, E_p recovers rapidly to the equilibrium locus but then overshoots before returning. E_f only starts to increase after E_p has overshot the equilibrium locus and this again indicates a lag in the response of the boundary layer to a change in pressure gradient. The failure of the boundary layer to return to the equilibrium locus far downstream (in region D) may be interpreted as a further indication of the persistently discreptive effect of the shock-wave boundary-layer interactions on the velocity profiles diready detected in the law of the wall as shown in Fig 21.

On the other hand the wake components of the profiles, remain fairly close to the standard shape as shown in Figs 23 and 24. In these figures the normalised wake components have been plotted in the following way. Fig 23a shows the undisturbed wake components in region A, and the results under the first compression through which it is possible to fit a law of the wall are shown in Fig 23b. Fig 24a-c has been produced by grouping similar normalised wakes into one diagram. In fact Fig 24a&b covers the rapid recovery region C together with the beginning of D, while Fig 24c covers the downstream part of region D.

According to Winter and Gaudet 25 the law of the wake for compressible boundary layers in zero pressure gradients is

$$\frac{\Delta U}{U_{\tau}^{i}} = 0.89 \left[1 + \sin \frac{\pi}{0.707} \left(\frac{y}{\delta_{0.999}} - 0.483 \right) \right] . \tag{39}$$

which after normalisation and allowance for the difference between $\delta_{0.999}$ and $\frac{1}{0.995}$ becomes

$$\left(\frac{\Delta U}{U_{\tau}^{i}}\right)_{N} = 0.5 \left[1 + \sin \frac{\pi}{0.818} \left(\frac{y}{\delta_{0.995}} - 0.529\right)\right] , \qquad (40)$$

where the normalised wake component $(\Delta U/U_{\tau}^{i})_{N}$ has been obtained by dividing the wake component by the maximum value for each traverse. The shape of the normalised wake components has been plotted in Figs 23 and 24. On each figure a curve in the form of equation (40) is shown with the constants adjusted to fit the particular set of results.

The maximum values of the wake component for each traverse $(\Delta U/U_{\tau}^{i})_{max}$ are plotted in Fig 25 against E_{f} . In Fig 25a the results for regions A and B ahead of the normal shock wave are given while the results downstream of the normal shock wave (in regions C and D) are shown in Fig 25b. It will be seen in Fig 25a that apart from the boundary layer profiles that have been perturbed by the leading edge of the strong compression, the maximum values of the wake components are in good agreement with the locus obtained from the equilibrium family of boundary layers described by East, Sawyer and Nash²⁸. However, in Fig 25b for locations downstream of the disruptive effect of the strong pressure gradient (points in the regions C and D downstream of the normal shock wave) the results depart considerably from the equilibrium locus.

The different behaviour upstream and downstream of the shock wave is illustrated further in Fig 26 where calculated values of J and K are plotted against the equilibrium parameter $E_{\rm f}$. The values of J and K were obtained by fitting sine waves of the form

$$\begin{pmatrix} \Delta U \\ U \\ \tau \\ \tau \end{pmatrix}_{N} = 0.5 \left[1 + \sin \frac{\pi}{J} \left(\frac{y}{\delta_{0.995}} - K \right) \right] , \qquad (41)$$

to the wake components for each boundary layer profile.

It should be noted that because of the quantity of data recorded, the values of $\left(\frac{\partial U}{\partial t} \right)_{max}$ were taken at that y-position of the pitot probe which gave a maximum value rather than estimating the true value by curve fitting. This in part accounts for the spread of values of J. No attempt was made to investigate the region B under the first compression.

However in the undisturbed equilibrium boundary layers of region A , the averaged values of J and K result in a normalised law of the wake

$$\left(\frac{\Delta U}{U_{\tau}^{i}}\right)_{N} = 0.5 \left[1 + \sin \frac{\pi}{0.819} \left(\frac{y}{\delta_{0.995}} - 0.494\right)\right]$$
(42)

which is extremely close to Winter and Gaudet's law (equation (40)). The average value of $\left(\Sigma U/U_{\pm}^{i} \right)_{max}$ for region A is 2.23 which is rather higher than Winter and Gaudet's value of 1.78 (equation (39)).

In the region downstream of the shock wave (Fig 26b) the parameters J and K show an appreciable variation with E_f . Thus in this region the velocity profiles cannot be represented by a family having either a standard law of the wall (section 4.5.3) or a standard form of the law of the wake.

5 CONCLUSIONS

Seven flows have been studied involving the interaction of normal shock waves with nominally two-dimensional turbulent boundary layers over a range of Mach numbers of 1.3 to 1.5 and of Reynolds numbers based on an effective streamwise run of 10×10^6 to 30×10^6 . The data extend from 700 mm upstream to 3000 mm downstream of the normal shock-wave position. These distances are equivalent to respectively 90 and 400 times the undisturbed boundary layer displacement thickness.

Oil flow investigations on the floor under the main interaction region indicate a highly three-dimensional flow at M = 1.5, but the results and momentum balance calculations support the view that this three-dimensionality is confined to the separation region. This region is very shallow having a depth of less than 10 mm and streamwise extent of 300 mm while the flow in the region of the tunnel centre line is tolerably two-dimensional, except in the immediate regions of the saddle points at the start and end of the separation. Further downstream of the main interaction region, the momentum balance calculations indicate a slight flow divergence over the width of the tunnel of between 7° at M = 1.3 to 15° at M = 1.5.

In calculating the momentum balance, use has been made of corrected static pressure measurements and an attempt has been made to follow the recent ideas of matching the tee-stream conditions with the viscous layer by defining an equivalent inviscid flow.

A method has been evolved for correcting the static pressure measurements made with the twin static pressure probe used in the experiment, and static pressure distributions formal to the wall within the main shock-wave boundary-layer interaction are presented.

The results agree with the laser measurements of East³ which cover a more limited \cdots . The present experiment has also produced more detail of the separated region. d agreement is also noted with Kooi's⁷ results at M = 1.4.

All the measurements confirm that the flow has not stabilised after 3 m downstream . The normal shock wave (or 400 times the undisturbed displacement thickness). Here the this shear layer which is produced at the bifurcation point of the shock waves at M = 1.4 with 1.5 is intact and there are still velocity gradients across the tunnel in the inviscid tiew. Also the boundary-layer profiles have not recovered to the equilibrium shapes.

The analysis in terms of the law of the wall and the law of the wake for the region whead of the interaction agree fairly closely with Winter and Gaudet (equation (29)) and less (equation (5)). However the evidence downstream of the normal shock wave suggests with the law of the wall has been changed by the large perturbations or normal pressure statients. However a single law of the wall may be used for the whole region behind the larm disturbance.

A normalised version of Winter and Gaudet's law of the wake (equation (4C)) provides () (air estimate for the shape of the normalised wake components ahead of the normal shock () (4). The disruptive effects of the normal shock waves appear mainly as charges in (mplifude of the normalised wake components, but the form of the wake component is also () (analoged.

Apart from the boundary layers immediately affected by the leading edge of the first mpression, the maximum values of the wake function ahead of the normal shock wave have similar correlation with the equilibrium function E_f (equation (36)) as the equilibrium function of flows of East, Sawyer and Nash²⁸. Although the maximum values of the wake functions downstream of the normal shock wave also correlate with E_f , they correlate in a quite different way from that expected for equilibrium boundary layers.

<u>Appendix</u> STATIC PRESSURE PROBE CORRECTIONS

A.1 Calibration measurements

As stated previously (section 2.2.4) the corrections to be applied to the static pressure probes were based on limited measurements made in the free stream at transonic speeds together with the results of the calibrations of the twin static pressure probe by traverses through the floor boundary layers at free-stream Mach numbers between 0.66 and 0.88. Additional information was derived during the actual experiment from the static and pitot traverses ahead of the interaction region at free-stream Mach numbers of 1.3, 1.4 and 1.5. Data was also used from the static and pitot traverses made through the boundary layer 1 to 3 m downstream of the normal shock wave at M = 1.3. It was assumed that for all these regions the static pressure for each traverse was constant and equal to the wall value, p_{tr} .

The results of the free-stream calibrations are shown in Fig 27 where the differences between the measured and actual static pressures are presented as pressure coefficients based on local conditions plotted against Mach number. Also (shown dotted) are the calibrations after three stages of smoothing which were used to correct the static pressure readings made under the severe velocity gradients of the region investigated (section A.2.2). There was no apparent Reynolds number effect on the calibrations but time limitations precluded much use of the single static pressure probes. The correction method described, therefore applies only to the twin static pressure probe. No static measurements were made at M = 1.4 for a unit Reynolds number of $3.5 \times 10^6/m$.

A.2 The correction method applied to the twin static pressure probe

Fig 28a&b shows typical static pressure measurement errors incurred while making boundary layer traverses with the twin static pressure probe. The boundary layers had constant static pressures normal to the wall.

The errors have been presented as the differences between the measured and actual static pressures and are shown as pressure coefficients based on local conditions. They are plotted against the distance from the wall, y, non-dimensionalised by dividing by the boundary layer thickness $\delta_{0.995}$.

Fig 28a shows typical errors when the maximum Mach number is less than the critical value of 1.098 (the point at which the free-stream calibration errors are shown in Fig 27, to become negligible). The errors have a different character if the maximum Mach number for the traverse is above the critical value as shown in Fig 28b. It is therefore convenient to divide the errors to be corrected into two categories:

- when the outer static pressure probe is at a Mach number of less than or equal to 1.098;
- (2) when the outer static pressure probe is at a Mach number greater than 1.098.

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This is a not unreasonable assumption because the main source of error in the treestream calibrations and the static traverse measurements made away from the influence of the wall, can be shown to be mostly due to the interconnecting stem between the inner and outer static pressure probes (see Fig 27 and Read. Pope and Cooksey²⁹, pp 1028103). Operatore the main correction to be applied to the inner static probe depends on some estimate of the mean Mach number over the connecting stem provided the static pressure probes are both at Mach numbers below the critical 1.098. If however, the outer static probe is at a Mach number above this critical value, then there is no interference consociated with the interconnecting stem in the region of the outer static probe. In this case a suitable course of action is to use the local Mach number to estimate the correction due to the interconnecting stem on the pressure measured by the inner static probe.

It is now proposed to deal with the errors in more detail by considering the three regions A, B and C indicated in Fig 28 in conjunction with the Mach number of the oter static pressure probe.

S. 2.1 Region A

The error in this region is treated as arising mainly from wall interference. There is however some compressibility effect. The errors are presented in law of the wall terms for correlation purposes (Figs 29 and 30). Fig 29 shows the accumulated errors resulting from the subsonic boundary-layer traverses, while Fig 30 shows the errors resulting from the undisturbed boundary-layer traverses ahead of the interaction region at tree-stream Mach numbers of 1.3, 1.4 and 1.5. To calculate the corrections, the errors region A may be added to those of region B. However because the errors of regions 6 and C are based on local conditions rather than the conditions at the edge of the boundary layer used for region A, the region A errors are multiplied by a factor $(1.50)^2$ to bring them into line.

Thus the corrections needed for the static pressure measurements made in region. A $\omega \in {\rm given}$ by



A.2.1.1 Outer static pressure probe at $M \le 1.098$

The measurement errors shown in Fig 29 have been presented as

$$\left(\frac{\mathbf{p}_{m}-\mathbf{p}_{w}}{\frac{1}{2}\boldsymbol{\rho}_{\beta}\boldsymbol{U}_{\delta}^{2}}\right)\frac{1}{C_{f}}$$

and plotted against yU_{τ}/v_{ξ} . The corrections needed are actually fairly small (for the Mach numbers are low) and so a single correction has been used and this is represented by the continuous line in the figure. The correction is given by

$$C_{p_{A}} = 15C_{f} \left(\frac{M_{\delta}}{M}\right)^{2} \qquad \text{when} \quad \frac{yU_{\pm}}{v_{\epsilon}} \le 644$$

$$C_{p_{A}} = C_{f} \left(\frac{M_{\delta}}{M}\right)^{2} \left[50.37 - 5.469 \ln \left(\frac{yU_{\pm}}{v_{\delta}}\right)\right] \qquad \text{when} \quad 644 \le \frac{yU_{\pm}}{v_{\epsilon}} \le 1000$$

$$C_{p_{A}} = 0 \qquad \text{when} \quad \frac{yU_{\pm}}{v_{\epsilon}} \ge 1000$$

A.2.1.2 Outer static pressure probe at M > 1.098

Fig 30 shows the pressure measurement errors plotted in similar terms to those used in section A.2.1.1. Here the results are strongly Mach-number dependent. The corrections have been applied on the assumption that the pressure measurement errors in Fig 30 have a linear relation with Mach number for constant yU_{χ}/ψ_{χ} . The slopes and zero intercepts of this linear variation are then assumed to be entirely dependent upon $yU_{\chi}/\psi_{\delta 0.995}$. The corrections which are shown by the continuous lines in Fig 30 are given by

$$C_{p_{A}} = FC_{f} \left(\frac{M_{\delta}}{M}\right)^{2} \left[M - 0.2725 - 0.1153 \ln\left(\frac{yU_{\tau}}{V_{f}}\right)\right]$$

where
$$F = 19.33 \ln\left(\frac{yU_{\tau}}{v_{\delta}}\right) - 433.5$$
 when $\frac{yU_{\tau}}{v_{\tau}} \le 7660$
 $F = 692.7 \ln\left(\frac{yU_{\tau}}{v_{\delta}}\right) - 6456$ when $7660 \le \frac{yU_{\tau}}{v_{\tau}} \le 11200$
 $F = 0$ when $\frac{yU_{\tau}}{v_{\tau}} \ge 11200$.

This correction is based on rather limited evidence but the magnitude is small and the probable accuracy of the correction is within $1\frac{1}{2}$ % of the true static pressure.

A.2.2 Regions B and C

The errors in these regions are treated as arising from a combination of shear flow and compressibility effects. It would therefore be expected that there would be an error which would correlate with $y/\delta_{0.995}$ and a larger error due to the geometry of the twin static pressure probe, which would correlate with Mach number. As stated in section A.2 the greater part of the latter error is due to the stem interconnecting the inner and outer static pressure probes, and the necessary correction depends on an estimate of Mach number and a form of the free-stream calibration of the twin static probe.

In fact, the average Mach number of the inner and outer static probes is used to give the necessary correction to the static pressure measured by the inner probe, when the outer probe is below the critical Mach number of 1.098. In all other instances the local Mach number is used to obtain the correction needed to the measured static pressure.

8.049

This applies to both the inner and outer static probes. The local Mach number is always used for the outer static pressure probe because it is normally in a region where the Mach number gradient normal to the wall is small.

It was found that the free-stream calibration overcorrected the static pressure measurement errors, and so a compromise calibration was used (shown dotted in Fig 27a) which involved three stages of smoothing. It may be noted from Fig 16a-f that the critical Mach number is only encountered ahead of and in the immediate vicinity of the main interaction region. The location of the point for critical Mach number is close to the wall for stations upstream of the influence of the normal shock wave. The environment is therefore one of severe shear strains and velocity gradients and the smoothing exercise is probably justified as it only affects the free-stream calibration in the region of the critical Mach number.

The form of the corrections is given by

$$C_{p_{B}} = fn\left(M_{I}, M_{U}, \frac{y}{\delta_{0.995}}\right)$$
$$C_{p_{C}} = fn\left(M_{U}, \frac{y}{\delta_{0.995}}\right) .$$

A.2.2.1 Outer static pressure probe at $M \le 1.098$

The corrections are calculated in terms of local density and velocity. The moditied calibration curves (shown dotted in Fig 27) are used to obtain $(p_m - p/\frac{1}{2} \circ U^2)$ from fn(M) and the modification to the free-stream calibration due to probe position is shown in Fig 31.

Thus the total correction is given by

$$C_{p_{B}} = fn\left(\frac{M_{I} + M_{U}}{2}\right) \times 0.94 \left[sin\left\{0.665\left(\frac{y}{\delta_{0.995}} - 0.15\right)\right\}\right]^{\frac{1}{2}}$$

$$C_{p_{C}} = fn(M_{U}) \times 0.94 \left[sin\left\{0.665\left(\frac{y}{\delta_{0.995}} - 0.15\right)\right\}\right]^{\frac{1}{2}}$$

$$C_{p_{B}} = 0 \qquad \text{if } \frac{y}{\delta_{0.995}} < 0.15 .$$

A.2.2.2 Outer static pressure probe at M > 1.098

$$C_{p_{B}} = fn(M_{I}) \times 0.94 \left[sin \left\{ 0.665 \left(\frac{y}{\delta_{0.995}} - 0.15 \right) \right\} \right]^{\frac{1}{2}}$$

$$C_{p_{C}} = fn(M_{U}) < 0.94 \left[sin \left\{ 0.665 \left(\frac{y}{0.995} - 0.15 \right) \right\} \right]^{\frac{1}{2}}$$
$$C_{p_{B}} = 0 \qquad \text{if} \quad \frac{y}{0.995} < 0.15 \quad .$$

A.2.3 Total corrections

Inner tube
$$C_{P_I} = C_{P_A} + C_{P_B} = \frac{P_m - p}{\frac{1}{2}cU^2}$$

Outer tube $C_{P_U} = C_{P_C} = \frac{P_m - p}{\frac{1}{2}cU^2}$.

A.3 Applying the corrections

The following describes how the static pressure corrections are calculated and applied:

- (1) Calculate the Mach number at each measuring station from interpolated pitot and wall pressure measurements. Calculate $y/s_{0.995}$, C_f and $y(U_f/v_s)$.
- (2) Calculate the static pressure corrections and subtract from the measured values.
- (3) Recalculate the new Mach number using the new static pressure and the interpolated pitot measurement.

Steps (2) and (3) are repeated until the difference between the new and old Mach number is less than 0.001. A suitable damping factor is needed to prevent oscillations in the calculation.

(4) Record the corrected Mach number and static pressure.

Fig 32 shows the effect of the correction on static pressures measured ahead and downstream of the interaction region where the static pressures may be expected to be nearly uniform across the boundary layer thickness. It will be seen that the static pressure errors have been reduced to $\frac{1}{2}$ % of the true static pressure.

A.4 Transferring the corrected static pressures to the pitot traverse measurements

The final stage of the procedure is to interpolate in the distribution of static pressure to obtain values at the location of the pitot-pressure measurements.

The calculated static pressures for each static pressure traverse are smoothed and transferred to a carpet plot of static pressure against streamwise position as shown in Fig 33. The appropriate static pressures are then transferred from the carpet plot for each pitot traverse. A final stage of smoothing ensures that the final static pressure curves are asymptotic to the wall static pressure within a few millimetres of the tunnel floor.

The corrected static pressure results may be seen in Fig 15a&b.

				_	_	_		_
Effective turbulent run ahead of wíndow centreline (m)	3.25	3.23	3.17	3.14	2.97	2.95	3.02	
Distance of throat ahead of window centreline (m)	2.78	2.78	2.84	2.84	2.86	2.87	2.87	
Re 2 × 10 ⁻⁶	11.2	31.2	10.9	29.5	9.5	17.3	27.9	
$\begin{array}{c} \text{Re} \ \theta \\ \times \ 10^{-3} \end{array}$	14.1	33.8	13.6	31.6	11.8	20.2	29.2	
Re &* × 10 ⁻³	29.1	68.2	30.0	68.2	28.1	46.1	68.7	
Re/ш × 10 ⁻⁶	3.66	10.21	3.67	10.03	3.51	6.47	96.6	
w	1.270	1.272	626.1	1.386	1.522	1.531	1.538	

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1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1	1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1.4     1		÷	68 83			1996 9	X 8 X X 8	8			1	CL/6 0		1522 0	51	1212		2		à		หั	67 1 6	<b>5</b> 0 0	52 	2 1 6 6 6		9228	
1.407     1.571     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574     1.574 <td< td=""><td>1.467     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     <td< td=""><td>1.400     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     <td< td=""><th>-</th><th>10 T 0</th><td>1</td><td>C SLUS</td><td>6 9929</td><td>8 2661 8 26</td><th>3</th><td>2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2</td><td>E 11 222</td><td>1</td><td></td><td></td><td>1007 B</td><td>69 <b>8</b></td><td><b>光</b>一</td><td></td><td><b>1</b></td><td></td><td>ð</td><td></td><td><b>3</b></td><td>53 - L</td><td>¥ 8,99</td><td>297 T 298</td><td>5 1 6 6</td><td>8 1713</td><td>ELVE 8</td><td></td></td<></td></td<></td></td<>	1.467     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476     1.476 <td< td=""><td>1.400     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     <td< td=""><th>-</th><th>10 T 0</th><td>1</td><td>C SLUS</td><td>6 9929</td><td>8 2661 8 26</td><th>3</th><td>2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2</td><td>E 11 222</td><td>1</td><td></td><td></td><td>1007 B</td><td>69 <b>8</b></td><td><b>光</b>一</td><td></td><td><b>1</b></td><td></td><td>ð</td><td></td><td><b>3</b></td><td>53 - L</td><td>¥ 8,99</td><td>297 T 298</td><td>5 1 6 6</td><td>8 1713</td><td>ELVE 8</td><td></td></td<></td></td<>	1.400     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800     1.800 <td< td=""><th>-</th><th>10 T 0</th><td>1</td><td>C SLUS</td><td>6 9929</td><td>8 2661 8 26</td><th>3</th><td>2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2</td><td>E 11 222</td><td>1</td><td></td><td></td><td>1007 B</td><td>69 <b>8</b></td><td><b>光</b>一</td><td></td><td><b>1</b></td><td></td><td>ð</td><td></td><td><b>3</b></td><td>53 - 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1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11	1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44     1.44	1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11	-	68 89	- 666	1 5497	8 9963	8 26 36 9 24	576	42 J 7 7	1.0001	4769	166 0	8/1/2	8.778	45 69	1 232	5766 8	1522	8 9926	8 218	8 248	æ	997 T 92	C(69) 20	32. 93.	6 1 8 8 8	9952 0 1		
1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     10000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000     1000	1.46       1.59       1.57       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.57       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56       1.56 <th< td=""><td>1.466     1.569     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     <th< td=""><th>-</th><th>12 B</th><td>-</td><td>1221</td><td></td><td>8 X44 8 X</td><th>614</th><td>26 27 1 25</td><td>8898 T (2)</td><td>26 -</td><td></td><td>8 273 1</td><td>9 2713</td><td>87. 178</td><td>1 248</td><td>286 9</td><td>1 2774</td><td>9266 9</td><td>2612 8</td><td>3492</td><td>88</td><td>2 1188</td><td>8968</td><td>5 1 31</td><td>7 1666</td><td>1 8 3336</td><td>9 3336</td><td></td></th<></td></th<>	1.466     1.569     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556     1.556 <th< td=""><th>-</th><th>12 B</th><td>-</td><td>1221</td><td></td><td>8 X44 8 X</td><th>614</th><td>26 27 1 25</td><td>8898 T (2)</td><td>26 -</td><td></td><td>8 273 1</td><td>9 2713</td><td>87. 178</td><td>1 248</td><td>286 9</td><td>1 2774</td><td>9266 9</td><td>2612 8</td><td>3492</td><td>88</td><td>2 1188</td><td>8968</td><td>5 1 31</td><td>7 1666</td><td>1 8 3336</td><td>9 3336</td><td></td></th<>	-	12 B	-	1221		8 X44 8 X	614	26 27 1 25	8898 T (2)	26 -		8 273 1	9 2713	87. 178	1 248	286 9	1 2774	9266 9	2612 8	3492	88	2 1188	8968	5 1 31	7 1666	1 8 3336	9 3336	
M3   <	1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1       1	1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.11     1.111     1.11     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.111     1.1111     1.1111     1.1111     1.1111 <th>-</th> <th>a a</th> <td></td> <td>1070</td> <td>866 8</td> <td>8 2614 8 26</td> <th></th> <td>52 19 1 25</td> <td>188 1 22</td> <td>1 4977</td> <td>1 691.</td> <td>8 2786 (</td> <td>9 2706</td> <td>55.69</td> <td>1 2338</td> <td>66.69</td> <td>1 3468</td> <td>1 6168</td> <td>5322</td> <td>6 3372</td> <td>87</td> <td>181 1 8</td> <td>10,000</td> <td>200</td> <td>6 1 666</td> <td>6 3002</td> <td>6 3627</td> <td></td>	-	a a		1070	866 8	8 2614 8 26		52 19 1 25	188 1 22	1 4977	1 691.	8 2786 (	9 2706	55.69	1 2338	66.69	1 3468	1 6168	5322	6 3372	87	181 1 8	10,000	200	6 1 666	6 3002	6 3627	
1.4   1.4   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2   1.2 <td>1.4.1       1.4.0       1.4.1       1.2.1       1.2.2       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       <th< td=""><td>1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1<th>• •</th><th></th><td></td><td>X</td><td></td><td>9 2614 8 26</td><th>11</th><td>100 11 120</td><td>23 0 9996</td><td>1 4976</td><td>1 6615</td><td>9 7,000</td><td>1572 B</td><td>162 11</td><td>1 224</td><td>9,9871</td><td>1 4928</td><td>1 1522</td><td>6175 8</td><td>6175 6</td><td>3</td><td>1 1 188</td><td>0.000</td><td>1 463</td><td>1 1 117</td><td>5 8 28H3</td><td>6 2843</td><td></td></td></th<></td>	1.4.1       1.4.0       1.4.1       1.2.1       1.2.2       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1       1.2.1 <th< td=""><td>1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   1   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## Table 3

# SUMMARY OF BOUNDARY-LAYER PARAMETERS

 $Re/m = 3.66 \times 10^{6}$  $M_0 = 1.270$ 

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룢	~ <b>8</b>	3.67	1 279	<b>4</b> ~	995 200	88	8	8 62	92 9	1 992	2 <b>84</b> 1	3.67	1.270	2.4	3 <b>269</b>	2 666	8	888	2 041	<b>4</b> ~	995 M	2 666	8
	۲ 88	50 20	1 267	2,206	3 507	2 078	1 339	8.944	92 9	198	2 663	<b>3</b> M	1 267	7. 286	3.507	2 <b>8</b> 78	1,338	<b>3</b>	5003	7.286	3.587	2 678	1 330
9,282-	8	3.67	1 261	382.2	<b>62</b> 2	5 861	123	8 635	98 9	2 865	2.111	3.67	1 261	2, 708	<b>82</b> 2 T	2.861	a	8	2 111	2, 708	<b>8</b> 2 Y	2 061	12
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89. 82	8 2j	388	L 215	8 979	₩ 378	82	1 366	622.7	<b>9</b> 2 '2	123	1.886	3 61	1.249	9.621	687 <del>-</del>	2 665	1.367	2.697	F 284	9.20	4.001	2 362	1.347
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57.8	ន័	32.2	L 6118 1	6 879	2.104	20	1 78	5 141	11 Se	8	<b>8</b> 644	3 <b>7</b> 2	8 987	16.989	2.678	2 <b>48</b>	1.819	5 14	0 661	16.989	7. 878	2.488	1.819
106. 0	ន័	387	1 994 1	7.921	7.551	R	1 815	12	12 23	8 679	0 764	<b>95</b> ~	8 361	17.948	2.549	2 378	1.821	2	9. 772	17.948	7.549	2 378	181
201.0	8	3 69 X	1 974 1	7.862	2.902.5	8	173	8	12 87	8	6, 962	3 49	8.974	17.862	7.967	2 259	62 J	<u>s</u>	8	17.862	286.2	2 239	1739
8	÷.	38 ₩	1 28	7.638	27 8	2	163	<b>8</b>	13.30	1.668	1.138	<b>8</b> 8. ∼	<b>6</b> 903	17.638	8 123	2 171	6	<b>4</b> 895	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	17 638	8 123	2.171	1.63
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<b>Gen</b>	E Bee	5,744	6.005	6 384	6.546	6 63	6.686	6, 788	6.992	7, 345	8.17	952 6	11. 208	13 399	15. 249	16.171	16.468	16.803	16.963	16. 748	16 473	16 309	16. 094	15 792	15 632	15.344	15.264	15. 487	2 2	15 283	15.229	15.010	15 072	15.406	15.437	15. 204	15.888	16.623 1	
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	₩.		1 282	1 267	1 266	1.272	1 272	1 273	134	L 249	1 219	1.181	1.002	1.614	88	8.964	8,963	8.955	8 3	8.938	8.934	0.93	8 928	8 922	6 919	<b>8</b> 913	8 987	86 96	8 8	88	<b>6</b> 8 <u>3</u>	<b>8</b> 892	8,887	<b>6</b> 875	<b>8</b> 73	8,866	6 856	8.80	
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	X10+3	1 818	1 766	1 733	1.771	1 78	1 783	1 747	1 694	1.551	1.316	1.047	<b>B.</b> 737	0.441	9 360	9 <u>2</u> 9	8.456	8 565	<b>8</b> 674	<b>6</b> . 748	8	6.874	8.922	1 641	1 692	1.262	1.316	1 28	1 334	1.33	1.444	1.54	1666	1.577	1 582	1.741	1 693	1.86	
	CFP X	1 81	1.814	186	<b>7</b>	1 82	1 866	1 73	1 731	156	1 376	1 101	8 814	6.456	<b>6</b> 33	8.410	<b>6</b> 487	0.641	<b>B</b> 695	698 B	<b>8</b> 878	6 961	1.015	1 111	1 289	1. 287	1 303	1 338	88	1.45	123	1567	1 64	1 732	1.686	178	1888	1 927	
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	86.∕₩ X10-6	3.64	3.61	3.64	367	3.67	3 <b>68</b> 7	365	361	<b>39</b> M	28	300	38	3	3.67	3	38	3 <b>9</b> %	3.63	<b>9</b> 8. ∼	3 <b>2</b> %	<u>ж</u>	<b>8</b> 5. ∼	3.43	<del>ور</del> ۲	3 33
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 $M_0 = 1.522$  Re/m = 3.51 × 10⁶

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8	Mereo	6 492	6,938	7.197	2.79	7.879	9,464	18. <del>4</del> 64	14, 889	17.459	2 22	27.756	<b>992</b> 22	<b>1</b> 8 <b>4</b>	43,643	45.69 <b>8</b>	<b>8</b> 23	18 ¥	<b>1</b> 2 62	82 X	37.272	688 X	ର୍ଷ ନ	<u>2</u> 12	38 471	29.005	28 965
	te te	1.98	1 885	1 866	1.865	163	1 662	8	0, <del>8</del> 66	<b>8</b> , <b>86</b>	8.608	<b>808</b>	B. 000	8 <b>66</b> 8	0 000 0	6 666	8 868	8 000	<b>9 000</b>	8, 211	6, 597	1. 697	1 23	1.593	1 670	174	1 639
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	<u>a</u>	1 278	1.206	1 286	蒸ぎ	1.346	1.491	<u>1</u> 83	1.918	2.196	5 5 7 7	2 978	3.176	3 88 7	<b>%</b> ~	<b>8</b> ~	3 <b>8</b> 6	3.476	315	5 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	184	156	143	1369	1 362	1.283	1.278
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	ġ₹	6 492	6 938	7. 128	200-2	2, 868	8 922	18 392	14.006	17.469	21 60	59 <b>19</b> 2	82 R	<b>1</b> 5 014	43.643	45.690	¥; 52	82.¥	42 62	45 755	37.272	33 <b>88</b> 9	<b>5</b> 20	31, 245	38.471	23. <b>BU</b> S	58 <b>3</b> 62
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	HI DELE CFP CF Mi X10+3 X10+3	9.799 4.97 1.944 1.929	9.439 5.31 1.869 1.865	9.716 5.46 1.854 1.888	9.268 5.68 1.776 1.612	9.044 5.83 1.648 1.697	7.989 6.22 8.925 1.648	6482 693 8.212 8.332	5.577 8.19 8.686 8.686	5,269 8,96 8,666 8,666	5,124 9.32 8,000 8,900	4,841 18,45 8,000 3,000	4 633 11 26 8 000 8 000	4 406 11 41 8 606 8 606	4,431,11,70,8,600,9,000	4,386, 12,52, 8,666, 8,666	4.528 13.12 8.888 8.888	4.179 15.29 0.000 0.000	4.211 16.33 0.000 0.000	4.254 28.94 8.218 8.211	4,749 25,68 8,648 8,597	5,554,27,92,8,996,1,667	6 609 29.31 1 236 1 233	6.634 J2.66 1.632 1.593	7.346 33.67 1.764 1.678	7.932 33.23 1.842 1.744	8174 33 79 1 888 1 839
	HBAR HI DELE CFP CF Man X10+3 X10+3	1 278 9.799 4.97 1 944 1 929	1286 9439 531 1869 1885	1.206 9.716 5.46 1.854 1.809	1296 9.268 5.68 1.776 1.642	1317 9.044 5.83 1.640 1.637	1.458 7.989 6.22 8.925 1.648	1.684 6.462 6.93 8.212 8.322	1.918 5.577 8.19 8.666 8.666	2.189 5.269 8.98 8.6666 8.666	2.614 5.124 9.32 8.008 9.908	3.648 4.841 10.45 6.666 3.666	3.389 4.639 11.25 8.669 8.669	4,259 4,468 11,41 0,000 0,000	4.618 4.431 11.79 8.000 8.000	4,448 4,386 12,52 8,666 8,666	4,282 4,528 13,12 8,888 8,888	3.466 4.179 15.29 8.666 8.666	3.204 4.211 16.33 9.609 9.609	2 466 4, 254 26, 54 9, 6, 218 8, 211	1. 814 4. 749 25. 68 8. 648 8. 597	1,563 5,554 27.52 8,996 1,667	1473 6.609 29.31 1.266 1.223	LX66 6.654 22.06 1.652 1.593	1 215 7.346 33.67 1.764 1.679	1.283 7.932 33.23 1.842 1.744	1 278 8 174 33 78 1 888 1 839
	H HBAR HI DELE GFP GF Mm X1043 X1043	2 364 1 278 9.799 4.97 1 944 1 959	2 359 1 286 9 439 5 31 1 869 1 865	2 358 1 236 2 716 5 46 1 854 1 898	2 223 1 234 9 268 5 68 1 776 1 612	2 283 1 317 9.044 5 83 1 640 1 697	2 249 1458 7.989 6.22 8.925 1.848	2 592 1.684 6.482 6.93 8.212 8.332	2863 1918 5.577 8.19 8.666 8.666	3.118 2.189 5.269 8.98 8.000 8.000	3 631 2 614 5 124 9 32 8 666 9 506	1. 144 3. 648 4. 841 16. 45 6. 666 3. 666	1,543 3,389 4,639 11,26 8,000 8,000	1 611 4.259 4.408 11.41 0.000 0.000	5,002 4,618 4,431 11.78 8,000 8,000	1771 4.448 4.306 12.52 8.000 8.000	1.522 4.262 4.528 13.12 8.888 8.888	1,529 3,406 4,179 15,29 8,000 8,000	1,144 3,284 4,211 16,33 9,666 9,666	X 100 2 400 4 254 28 34 9 238 9 211	2 225 1 814 4 749 25 68 8 648 8 597	2.01.2.1.503 5.554 27.52 0.996 1.007	1864 1473 6 609 29.31 1286 1233	L706 1369 6.654 22 66 1.652 1593	L 629 L 215 7.346 33 67 1 764 1 678	1572 1283 7.922 33.23 1842 1744	1552 1279 8174 3378 1888 1839
CONSTRAFT PN	Thethe H Hande HI Delle (JP (J Mi Xulm3 Xulm3 Xulm3 Xulm3	2746 2364 1278 9.799 4.97 1.944 1.959	2 941 2 359 1 266 9 439 5 31 1 869 1 865	3 623 2 359 1 236 9 716 5 46 1 854 1 869	3156 2 29 1 296 9 268 5 68 1 776 1 812	3 266 2 283 1 317 9.044 5 83 1 640 1 697	3 663 2 249 1 459 7 989 6 22 8 925 1 648	4.122 2.582 1.684 6.482 6.93 8.212 8.322	5 666 2 863 1 948 5 577 8 19 8 666 8 666	5 592 3 118 2 189 5 269 8 96 8 966 8 666	5 554 3 651 2 614 5 124 9 22 8 666 8 566	6.744 4.144 3.048 4.841 18.45 8.000 3.000	7.341 4.543 3.309 4.639 11.26 8.000 8.009	7.276 5.641 4.259 4.408 14.41 8.608 8.609	7.452 6.023 4.618 4.431 11.79 8.000 8.000	8 883 5 771 4,448 4,386 12.52 8,888 8,889	8 576 5 532 4 262 4 529 13 12 8 888 8 669	10, 346, 4, 529, 3, 406, 4, 179, 15, 29, 0, 000, 9, 000	14 642 4,144 3,264 4,211 16,33 8,866 9,669	13.788 3.184 2.488 4.254 28.94 8.218 8.211	16 838 2.325 1.814 4.749 25.68 8.648 8.597	16.846 2.642 1.583 5.554 27.52 0.996 1.667	7.22 1664 1473 6.609 29.31 1266 1233	18.312 1.766 1.369 6.654 22 66 1.652 1.593	18.789 1.629 1.345 7.346 33.67 1.764 1.678	8 454 1 572 1 283 7.932 33 23 1 842 1 744	8.661 1.552 1.278 8.174 33.78 1.888 1.839
CONSTRAFT PN	DEL+ THETH H HERR HT DELE OFP OF 1911 1911 - 1943 XLI0+3 XLI0+3	6492 2746 2364 1278 9.799 4.97 1.944 1.959	6 338 2 941 2 359 1 266 9 439 5 31 1 869 1 865	7.128 3.023 2.358 1.236 9.716 5.46 1.854 1.889	7.349 3.156 2.229 1.256 9.268 5.68 1.776 1.812	7.457 3.266 2.283 1.317 9.044 5.83 1.646 1.697	8.227 3.663 2.249 1.459 7.999 6.22 8.925 1.648	18 313 4.122 2.592 1.694 6.402 6.93 8.212 8.322	[4, B15 5, B69 2, 883 1, 918 5, 577 8, 19 8, 699 8, 699	17.434 5.552 3.118 2.189 5.269 8.98 8.000 8.000	21 666 5 554 3 651 2 614 5 124 9 22 8 666 9 566	27.952 6.744 4.144 3.048 4.841 10.45 6.000 3.000	CC 348 2 341 4 543 3 389 4 609 11 26 8 666 8 666	16, 1228 7, 276 5, 611 4, 259 4, 406 11, 41 8, 606 8, 609	M 882 7.452 6.823 4.648 4.431 11.79 8.666 8.669	16, 649 8, 883 5, 771 4, 448 4, 306 12, 52 8, 688 8, 609	17.442 8.576 5.552 4.262 4.529 13.12 8.888 8.889	NG 767 140 346 4.529 3.496 4.179 15.29 8.000 8.000	15 628 11 642 4 144 3 284 4 211 16 33 8 666 8 668	12 753 13 788 3 104 2 408 4 254 28 94 8 218 8 211	77. 272 16 609 2. 225 1. 614 4. 749 25. 68 8. 648 8. 597	03 889 16 846 2 612 1.563 5 554 27.52 8 996 1 667	2 294 17.222 1664 1473 6 609 29.31 1286 1233	11.245 18.312 1.766 1.369 6.654 22 66 1.652 1.593	00 472 18 789 1 629 1 115 7. 346 33 67 1 764 1 678	29.045 18.454 1.572 1.283 7.932 33.23 1.842 1.744	<b>26 965 18 661 1 352 1 278 8 174 33 78 1 888 1 839</b>
CONSTRAIT PN	FE DEL: THETH H HORP. HI DELE CFP CF FM FM FM HORP. HI X10H3 X10H3	L3M 6.482 2.746 2.364 1.278 9.799 4.97 1.944 1.559	1522 6 938 2 941 2 559 1 266 9 439 5 31 1 669 1 665	1534 7.128 3.023 2.359 1.236 9.716 5.46 1.854 1.889	LSH 7.349 3156 2329 1296 9.268 5.68 1776 1612	L 461 7.467 3.266 2.283 1.317 9.004 5.83 1.640 1.697	1223 8227 3663 2249 14549 7.949 6.22 8.925 1648	L 244 18 313 4,122 2 592 1 684 6 462 6 53 8 212 8 332	L 229 14 645 5 666 2 863 1 918 5 577 8 19 8 666 8 669	L 2017 17.434 5.552 3.118 2.169 5.269 8.98 8.000 8.000	L106 21.606 5.954 3.631 2.614 5.124 9.22 8.608 9.906	L 164 27.952 6.744 4.144 3.048 4.041 10.45 6.000 3.000	L 152 33 348 7 341 4 543 3 389 4 639 11 26 8 669 8 669	. 134 48 828 7. 276 5. 611 4. 259 4. 408 11. 41 8. 608 8. 609	L 118 44.882 7.452 6.803 4.618 4.431 11.78 8.686 8.686	182 46 649 8 683 5 771 4 448 4 386 12 52 8 668 8 669	. 008 47.442 8.576 5.552 4.262 4.529 13.12 8.008 8.009	. 674 46.767 18.346 4.529 3.466 4.179 15.29 8.666 8.666	1 669 45 628 11 642 4 144 3 264 4 211 16 33 8 666 9 666	1 888 42 753 13 788 3 184 2 468 4 254 28 94 238 9 21	1 953 37.272 16 609 2.225 1.614 4.749 25.68 8.648 8.597	1911 33 889 16 846 2 642 1 543 5 554 27 92 8 996 1 667	1889 2 294 17 22 1 864 1 473 6 609 29 31 1 256 1 233	1856 31.245 18.312 1.766 1.369 6.654 32 66 1.652 1.593	1823 38 472 18 789 1629 1 XI5 7.346 33 67 1 764 1 676	1706 29:005 18:454 1.572 1.283 7.932 33.23 1.842 1.744	1788 28.965 18.661 1552 1.278 8.174 33.78 1.888 1.839
CONSTRAIT PN	NGAM NG DELA THETA N HBAR HI DELE OFP OF 19 <del>-</del> 6 na na haa hBar hi Xuda3 Xuda3	645 1544 6482 2746 2364 1278 9.799 4.97 1944 1359	645 1572 6938 2941 2559 1266 9439 531 1669 1865	6 47 1 551 7 128 3 623 2 558 1 256 9 716 5 46 1 854 1 889	649 1584 7.349 3156 229 1296 9.268 548 1776 1612	645 1461 7457 3266 2283 1317 9.044 583 1649 1697	6.29 1.22 8.227 3.663 2.249 1.458 7.999 6.22 8.255 1.648	6.58 1.2M 18.313 4.122 2.582 1.684 6.482 6.93 8.212 8.322	6 66 1 239 14 645 5 669 2 883 1 948 5 577 8 19 6 666 8 669	6.66 1.287 17.434 5.592 3.118 2.189 5.269 8.99 8.000 8.000	6.69 1.166 21.666 5.561 3.631 2.614 5.124 9.22 8.666 8.566	672 1 164 27.952 6 744 4 144 3 648 4 841 16 45 6 666 3 666	6 67 1 152 33 348 2 341 4 543 3 388 4 659 11 26 8 669 8 669	6 66 1 134 48 828 7.276 5 611 4.259 4.468 11 41 8.669 8.669	6.64 1.118 44.882 7.452 6.823 4.618 4.431 11.78 8.886 8.889	6 64 1 182 46 649 8 883 5 771 4 448 4 306 12 52 8 888 8 869	6.62 1.668 47.442 8.576 5.552 4.262 4.529 13.12 8.668 8.669	6.63 1.674 46.767 18.346 4.529 3.466 4.179 15.29 8.666 8.666	6 62 1 669 45 628 11 642 4 144 3 264 4 211 16 33 8 666 8 666	6 55 1 666 42 753 13 786 3 164 2 466 4 254 26 9 9 238 9 211	6 46 8 963 37.272 16 809 2.225 1.814 4.749 25.68 8.648 8.597	6.38 8.941 33.869 16.846 2.802 1.563 5.354 27.52 8.999 1.867	6.26 8.889 2.294 17.22 1.864 1.473 6.609 29.31 1.286 1.233	6 88 8 856 31 245 18 312 1766 1369 6 654 22 66 1652 1593	5.93 8.823 38.472 18.789 1.629 1.315 7.346 33.67 1.764 1.678	5.77 8.796 29.605 18.454 1.572 1.283 7.952 33.23 1.802 1.744	5.63 8.788 28.965 18.661 1.552 1.278 8.174 33.78 1.888 1.839
CONSTRAFT PN	L 955 RE/M RE DRIL® THETR H HBHR HI DRILE (JP (JF ) 18 X18+6 19 19 19 19 19 19 19 X16+3 X16+3	334 642 1544 6462 2746 2354 1278 9.799 4.97 1.944 1.559	34.7 6.45 1522 6.938 2.941 2.559 1.266 9.439 5.31 1.669 1.865	36.5 6 47 1531 7.128 3.023 2.358 1.256 9.716 5.46 1.854 1.888	X6 649 1394 7.349 3156 229 1294 9.268 548 176 1812	77.8 6.45 1.461 7.457 3.266 2.283 1.317 9.044 5.82 1.648 1.697	27.5 6.29 1.223 8.227 3.663 2.249 1.459 7.959 6.22 8.925 1.648	36.7 6.59 1.2M 18.313 4.122 2.582 1.694 6.462 6.53 8.212 8.322	41.9 6 66 1 229 14 605 5 666 2 683 1 918 5 577 8 19 8 666 8 669	46.9 6.66 1.207 17.434 5.552 3.118 2.189 5.269 8.99 8.000 8.000	S21 6.09 1106 21.606 5.954 3.631 2.614 5.124 9.22 8.608 8.908	GB 6 6 72 1 164 27.952 6 744 4 144 3 048 4 041 18 45 8 000 3 000	G.4 6 G 1152 33.348 7.341 4.543 3.388 4.639 11.26 8.669 8.669	72 9 6 66 1 134 48 628 7.276 5 611 4.259 4.468 11 41 8.669 8.669	77.9 6.64 1.118 44.882 7.452 6.823 4.618 4.421 11.78 8.686 8.669	82.1 6.64 1.182 46.649 8.883 5.771 4.448 4.386 12.52 8.666 8.669	6K.2 6.62 1.668 47.442 8.576 5.552 4.282 4.529 13.12 8.669 8.669	99.0 6.63 1.674 46.767 1.0.346 4.529 3.466 4.179 15.29 8.6669 8.669	92.0 6.02 1.059 45.628 11.012 4.144 3.204 4.211 16.33 8.000 0.000	1044 6.55 1 0006 42 753 13 768 3 104 2 408 4 254 26 9 0 238 0 21	113.4 6.46 8.963 37.272 16.639 2.225 1.614 4.749 25.68 8.648 8.597	127.4 6.38 8.911 33.889 16.846 2.612 1.543 5.551 27.92 8.996 1.667	136.9 6.36 8.889 2.294 17.222 1.864 1.473 6.869 29.31 1.286 1.233	153.1 6.08 8.056 31.245 18.332 1.746 1.369 6.654 22.06 1.652 1.593	167.9 5.93 8.823 38.472 18.789 1.629 1.115 7.346 33.67 1.764 1.679	17.4 5.7 8.796 29.005 18.454 1.572 1.283 7.932 33.23 1.842 1.744	1811 5 5 63 8 788 28 965 18 661 1 552 1 278 8 174 33 78 1 888 1 839

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Table 3 (concluded)

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 $M_0 = 1.538$  Re/m = 9.96 × 10⁶

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<b>4</b> 66. <b>B</b>	<u> 3</u> 3.3	<u>9</u> 91	1 54	6 270	2 664	<b>5</b> 7	1.261	18.146	<del>4</del> 88	1.924	. 836	9.91	154	6. 270	2 664	2 204	1 261	8 146	ŝ	6.278	2 664	28	1 261
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-169.8	36.1	<u>9</u> 81	1 288	2, 886	3 549	5 22	1.454	2.945	6.13	6 777	<b>8</b> . 866	<b>3</b> 6	1.428	8 277	3,669	2 20	1. <del>1</del> 86	195.7	8 833	8.159	3.809	2 123	1.449
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-12 0	2.8	18.42	1.218	13 884	4.878	2.851	1.978	5.896	<b>%</b> ~	6 666 966	9 609 9 609	19,48	1.272	198 ∑	4,969	2 803	L 979	5.051	999 999	13 831	5.047	2.748	199
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9.12	28.3	<b>10</b> 33	111	2, 97	6 161	3.557	2 576	206	<u> </u>	<b>8 666</b>	8 868 9	16.37	1.198	27.749	6 241		2 534	5.867	999 1990	21 736	6.277	ŝ	2 549
-24.8	6,5	10.28	1.148	N 62	6.545	4.526	8	5 229	10.22	6669	<b>8</b> . 899	10 30	1 155	19 19 19 19	6/2 3	8	3.217	5,207	9 <b>99</b> 0	29.275	6.637	444	3.210
8 92 72	9 <del>(</del> 9	<b>10</b> , 28	1121	296 XZ	7. 666	5. 897	3 873	4, 767	16 83	<b>9 900</b>	<b>8</b> . 888	10.17	1 661	Q 172	<b>585</b> '2	1 637	3.474	4 533	88	X 172	2882.7	4 637	3.474
125.0	88	<b>10</b> 23	1 (69	42 251	8 372	5.659	3886	4 258	12 71	9 900 1	8 8 9 9 9 9 9	18.15	1.047	15 338	8.378	2000	3.879	4.261	8 8 8 9 9 9	42 338	8.370	5.608	3.879
8	8	18, 17	1 663	41 932	18,464	<b>4</b> . 887	3. <b>68</b>	3.818	15.59	8.666	<b>6</b> . 8669	16.18	1 681	17 933	18 <del>4</del> 62	88 •	3 878	3 811	<b>800</b>	41 933	18 462	<del>4</del> 88	3.678
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415.9	8	18 08	1 012	102.02	12 412	3 206	2. <b>49</b> 8	4, 238	18.79	8.183	0, 175	10 05	1 683	5. 5. 5.	12 413	3 <b>2</b> 80	2 <b>4</b> 94	ង្កី	<b>8</b> . 176	<b>2</b> . 9	12 413	3,206	2 491
514.0	<u>8</u> 2	10.13	88 88 8	37 924	13 172	2 879	2.245	4 371	<b>58</b> 53	8 245	<b>8</b> 233	10, 12	 388 98	32.928	LI 172	5.829	2.245	4 371	20	37.928	13, 172	2.879	2,245
612.9	99.1	566	<b>8</b> 969	36. 716	14. 142	2.596	2.608	4.411	22.14	<del>8</del> 376	<u>8.461</u>	6.6		86 716	14. 142	528 538	5 <b>6</b> 08	4.411	8 19	36.716	14, 142	2,596	2.628
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9 888 9 888	107.5	<b>8</b>	8 937	33 757	14.995	2 251	1.766	4. 318	24, 18	8 661	0.641	8 8	6 937	122	14 995	5 291	7 766	4. 918	8. GA1	22 X2	14.995	2 231	1 766
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2317 8	15A.6	9 IJ	8.817	23 019	17.478	1 693	1.237	7 246	31.11	1 783	1 621	9 13	8.817	28 010	17 478	1 660	۲ <u>ک</u>	2 246	1.621	28.019	17 478	1 603	1 297
2708.8	162.8	862	8 792	S82 93	17 214	1.556	1 271	<b>19</b> 2007 - 2007	38.74	1 204	1 824	862	8 792	682 92 92	17 214	1.55	L 271	58.2	1.824	<b>682</b> 92	17 24	1 556	1 271
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### LIST OF SYMBOLS

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		(Bracketed symbols represent computer headings)
A,B		'law of the wall' constants in equation (28)
$^{C}{}_{\mathbf{f}}$	(CF)	local skin friction coefficient obtained from East's prediction of the law of the wall; see equation (6)
^C fp	(CFP)	local skin friction coefficient based on Patel's Preston tube calibration as formulated by Head and Vasanta Ram; see section 3.3
С _р		pressure coefficient based on local conditions
d		probe diameter
D		constant in equation (6)
E.		equilibrium factor in equation (36)
^Е р		equilibrium factor in equation (37)
F		compressibility factor in equation (6)
G		Clauser shape parameter in equation (32)
h		distance of probe centre line above wall
н	(H)	boundary layer shape parameter in equations (9) and (13)
Ĥ	(HBAR)	boundary layer $\varepsilon^{*}$ is a parameter in equations (10), (14) and (20)
н _I	(HI)	boundary layer shape parameter $\frac{\delta_{0.995} - \delta^*}{\theta}$
J,K		normalised 'law of the wake' constants in equation (41)
1		effective turbulent run ahead of normal part of shock wave
м	(M)	Mach number
	(ME)	Mach number at edge of boundary layer
P	(P)	static pressure
	(PI)	equivalent inviscid static pressure
Pt0	(PTO)	tunnel total pressure
Ptl		local total pressure
Re	(RE)	unit Reynolds number; see equation (3)
Т		temperature in Kelvins
	(TO)	total temperature in Kelvins
U		horizontal component of velocity
	(UE)	horizontal component of velocity at edge of boundary layer
ΔU		wake component of velocity profile
υ _τ		friction velocity

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### LIST OF SYMBOLS (concluded)

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$v^i_{\tau}$		equivalent incompressible friction velocity; see equation (26)
v		vertical component of velocity
х		streamwise position relative co the normal part of the main shock wave
У	(Y)	equivalent height above wall; see equations (1) and (2)
у*		non-dimensionalised y with respect to wall parameters in equation (6)
Z		distance perpendicular to flow but parallel to surface
δ		boundary layer thickness
⁸ 0.995	(DEL995)	boundary layer thickness at $U/U_{\delta} = 0.995$
^č 0.999		boundary layer thickness at $U/U_{\hat{\Delta}}$ = 0.999
ÊE	(DELE)	boundary layer energy thickness; see equation (15)
δ	(DELBAR)	bounda.y layer thickness parameters; see equation (7)
5 <b>*</b>	(DEL*)	boundary layer displacement thickness; see equations (11) and $(18)$
θ	(THETA)	boundary layer momentum thickness; see equations (12) and $\psi_{i}$
8	(THE TABAR)	boundary layer thickness parameter; see equation (8)
κ		mixing length constant in equation (27)
×.		surface curvature
5		viscosity of fluid
a.		kinematic viscosity of fluid
11		pressure gradient parameter for equilibrium flows; see equation $\mathbb{C}^{n}$
		density

### Subscripts

i	equivalent inviscid flow quantities
1	inner static pressure probe
m	measured quaraities
N	normalised quantities
0	tree-stream conditions before start of interaction
р	intermediate equivalent inviscid flow quantities
ť	outer static pressure probe
w	wall conditions
А,В,С	regions in Fig 28a&b
ŝ	edge of boundary layer conditions

Superscript

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equivalent incompressible flow quantities

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MICROCOPY RESOLUTION TEST CHART National Bureau of Standards - 1963 - 2

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Fig 1a-c Three techniques for producing a normal shock-wave boundary-layer interaction

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Fig 2

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Fig 3 False floor and horizontal traverse mechanism

Fig 3



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Fig 4 Pitot traverse mechanism and twin pitot probe

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Fig 4



Fig 5 Details of static probes

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Fig 6a







 $M_0 = 1.27$ Re/m = 10.2 x 10⁶

M₀ = 1.39 Re/m = 10.0 x 10⁶



 $M_0 = 1.54$ Re/m = 9.96 x 10⁶

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Fig 6b Schlieren photographs of normal shockwave boundary-layer interaction



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Fig 7

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Strong convergence lines

Saddle points

Separation node



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Fig 8

Fig 9



Fig.9 Momentum balance based on measured  $\,p\,$  + calculated  $\,p_{j}\,$  for  $\,X\,<\,0\,$ 

Change of scale 561 50 50 500 50 50 • Re/m = 3.66 × 10⁶  $\frac{d\phi}{dz}$  deg/m $Re/m = 10.2 \times 10^{6}$ + 40 400-40 40 40 30 30 30 300 30 20 20 - 200 20 20 0 10 10 10 100 10 ø + °o + 0 o 0 Ω 1000+ 2000 Xmm Б 0 + + 0 + +o -10 -10 -10 -10 + Θ -20 -20 -20 -20 + -30 -30 -300 -30 -30 -40 -40 -400 -40 -40 - 50 - 50 - 50 -500 -50



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Fig 10a

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Fig 10b



Fig 10b Flow divergence needed to balance momentum integral equation, M = 1.4

Fig 10c

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Fig 11



Fig 12 Composition of right hand side of momentum integral equation (measured  $p + calculated p_i$ )

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Fig 12

Fig 13a



Fig 13a Boundary layer development ( $p = p_w$ ) M = 1.3

Fig 13a (concld)





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Fig 13b





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Fig 13b (concld)



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Fig 13c



Fig 13c Boundary layer development (p =  $p_W$ ) M = 1.5

Fig 13c (concld)



## Fig 13c (concld) Boundary layer development (p = $p_w$ ) M = 1.5

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Fig 14a

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Fig 14a Boundary layer development (measured p) M = 1.3



Fig 14b

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Fig 14c

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Fig 14c Boundary layer development (measured p)  $M \approx 1.5$ 



Fig 15a Measured static pressures

Fig 15a

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Fig 15b

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## Fig 15b Measured static pressures

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Fig 16a

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Fig 16b

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Fig 16b Mach number distribution,  $M_0 = 1.27$ , Re/m = 10.2 x 10⁶



Fig 16c Mach number distribution,  $M_0 = 1.39$ , Re/m = 10.0 x 10⁶

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Fig 16c

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Fig 16d

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Fig 16d Mach number distribution,  $M_0 = 1.52$ ,  $Re/m = 3.51 \times 10^6$ 

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Fig 16e

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Fig 16f

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Fig 17a Velocity profiles,  $M_0 = 1.27$ , Re/m = 3.66 x 10⁶

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Fig 17a

Fig 17b





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Fig 17c Velocity profiles,  $M_0 = 1.39$ , Re/m = 10.0 x 10⁶

Fig 17c

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Fig 17d



Fig 17d Velocity profiles,  $M_0 = 1.52$ , Re/m =  $3.51 \times 10^6$ 

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Fig 17e

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Fig 17f

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Fig 18 Logarithmic velocity profiles ahead of interaction (region A) (see Fig 22)

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Fig 19 Logarithmic velocity profiles approx 3 m downstream of normal shock-wave (region D) (see Fig 22)

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Fig 19



Fig 20 Logarithmic velocity profiles just downstream of normal shock-wave (region C) (see Fig 22)

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Fig 20






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Fig 23a&b



Fig 23a&b Wake components of velocity profiles ahead of shock-wave

Fig 24a&b



Fig 24a&b Wake components of velocity profiles downstream of shock-wave

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## Fig 24c Wake components of velocity profiles downstream of shock-wave

Fig 24c

Fig 25a&b



Fig 25a&b Maximum values of wake component

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Fig 26a&b

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(b) Downstream of normal shockwave

Fig 26a&b Constants J and K in  $(\Delta U/U_{\tau}^{i})_{N} = 0.5(1 + \sin \pi/J(y/\delta_{0.995} - K))$ 

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Fig 27a&b



Fig 27a&b Free stream calibration of static probes

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Fig 28a&b

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Fig 28a&b Typical static pressure measurement errors





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## Fig 30 Wall/static interference errors, $M_{\mu} \ge 1.098$

Fig 30



Fig 31 Modification to smoothed free stream static pressure calibration caused by probe resition

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Fig 31

Fig 32a&b

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Fig 32a&b Typical corrected static pressures



Fig 33 Smoothed static pressure distribution, M = 1.5, Re =  $10 \times 10^6$ /mm

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## **REPORT DOCUMENTATION PAGE**

Overall security classification of this page

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This Report printeraction between layer. The measurem and at Reynolds num $30 \times 10^6$ . The result and static probes.	resents the results of a normal shock wave a ments were made at fre bers based on an effect its were obtained from	a study of and a two-di e-stream Ma tive stream a comprehens	seven flow Mensional a ch numbers wise run of ive travers	involving the surbulent boundary of 1.3, 1.4 and 1.5 $10 \times 10^6$ to be with both pitot
Standard bound static pressures are number distribution	ary-layer integral provident of the second s	with veloci region.	sed on wall ty profiles	and measured and the Mach
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